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Der Präsident des Europäischen Patentamts;
Im Auftrag

For the President of the European Patent Office

Le Président de l'Office européen des brevets
p.o.

I.L.C. HATTEN-HECKMAN



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(Falls die Bezeichnung der Erfindung nicht angegeben ist, siehe Beschreibung.
If no title is shown please refer to the description.
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Ranging system for determining ranging information of a spacecraft

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COMMUNICATION

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As requested, the entries pertaining to the applicant of the above-men-
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Société Européenne des Satellites S.A.

**Ranging System for Determining Ranging Information of a
Spacecraft**

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The invention refers to a ranging system and a ranging method for determining ranging information of a spacecraft carrying a component of a communication channel.

15

Furthermore, the invention refers to a receiving station and a method for operating a receiving station, a processing station and a processing method, a spacecraft and a method for operating a spacecraft, and a transmission unit and a method for operating a transmission unit.

20

Geo-stationary communication satellites have to be positioned in precisely defined space segments assigned by the International Telecommunication Union (ITU). However, the combined effect of oscillations of a period of 24 hours due to the inclination and eccentricity and the long-term drift of the main longitude leads to an apparent movement of the satellite with respect to its nominal position.

25

Therefore, the satellite has to be controlled under the effect of these perturbations by applying periodic orbit corrections in the most economic manner so that the satellite remains within the so-called station-keeping box. For such a control it is necessary to determine precisely the position and/or movement of the satellite which is usually addressed as ranging.

30

Due to sophisticated orbit positioning techniques of geo-stationary satellites, the requirements for the accuracy of ranging have increased recently. From DE 198 36 602 A1 it is known to position satellites in a so-called satellite cluster at one geo-stationary orbit position in order to make better use of the narrow orbit space. This means that not just one satellite, but a plurality of satellites have to be controlled in one station-keeping box.

The basic technique for determining the space coordinates of a satellite is based on a trilateration solution as described below. Fig. 1 shows the basic configuration of a trilateration measurement. Three transmission stations 101, 102 and 103 are located on earth, wherein the positions P1, P2 and P3 of each transmission station are known. If the distances d1, d2 and d3 can be measured, then the unknown position P4 of the satellite 104 can be determined. Let d_i denote the distance of the corresponding measurement of each transmission station and let (x, y, z) and (x_i, y_i, z_i) denote the Cartesian coordinates of the satellite 104 and each transmission station P_i, respectively. Then the following relation holds:

$$d_i = \sqrt{(x - x_i)^2 + (y - y_i)^2 + (z - z_i)^2} \\ - \sqrt{(x - x_0)^2 + (y - y_0)^2 + (z - z_0)^2} = f_i(\vec{q}), \quad i = 1, 2, 3 \quad (1)$$

where $\vec{q} = [x, y, z]^T$ is the unknown position vector of the satellite 104. The vector of the distance measurements is expressed as:

$$\vec{d} = \vec{f}(\vec{q}) \quad (2)$$

A commonly employed method to solve for q in this non-linear equation is the Gauß-Newton iterative method. The best estimate of \vec{q} is iteratively approximated as:

$$\hat{\vec{q}}_{k+1} = \hat{\vec{q}}_k + \left(\vec{F}^T \vec{F} \right)^{-1} \vec{F}^T \left(\vec{d} - \vec{f} \left(\hat{\vec{q}}_k \right) \right) \quad (3)$$

where \vec{F} is the Jacobian matrix:

$$\vec{F} = \frac{\partial \vec{f}}{\partial \vec{q}} = \begin{bmatrix} \frac{\partial f_1}{\partial x} & \frac{\partial f_1}{\partial y} & \frac{\partial f_1}{\partial z} \\ \frac{\partial f_2}{\partial x} & \frac{\partial f_2}{\partial y} & \frac{\partial f_2}{\partial z} \\ \frac{\partial f_3}{\partial x} & \frac{\partial f_3}{\partial y} & \frac{\partial f_3}{\partial z} \end{bmatrix} \quad (4)$$

10 However, in practice not only a trilateration configuration according to Fig. 1, but any other configuration like e.g. a bilateration or a quadrilateration configuration can be used.

15 Although a bilateration does not yield the full ranging information of a spacecraft, a bilateration still might be useful if only a certain coordinate of the spacecraft is of interest.

20 Furthermore, a quadrilateration configuration allows the solution for an additional unknown quantity. Fig. 2 shows the practical configuration of a quadrilateration measurement. Let say the transponder delay D of the transponder carried by the satellite is an additional
25 unknown quantity besides the space coordinates of the spacecraft. Typically, such a transponder comprises frequency converters, high-power amplifiers etc. Then the approach for each distance measurement d_i is as follows:

$$\begin{aligned}
 d_i &= D + \sqrt{(x - x_i)^2 + (y - y_i)^2 + (z - z_i)^2} \\
 &\quad - \sqrt{(x - x_0)^2 + (y - y_0)^2 + (z - z_0)^2} \\
 &= f_i(\vec{q}), \quad i = 1, 2, 3, 4
 \end{aligned} \tag{5}$$

where $\vec{q} = [x, y, z]^T$ is again the unknown position vector of the satellite. The solution for \vec{q} can be found by rewriting the formulas (2), (3) and (4) accordingly. The Jacobian matrix is then:

$$\vec{F} = \frac{\partial \vec{f}}{\partial \vec{q}} = \begin{bmatrix} \frac{\partial f_1}{\partial x} & \frac{\partial f_1}{\partial y} & \frac{\partial f_1}{\partial z} & \frac{\partial f_1}{\partial D} \\ \frac{\partial f_2}{\partial x} & \frac{\partial f_2}{\partial y} & \frac{\partial f_2}{\partial z} & \frac{\partial f_2}{\partial D} \\ \frac{\partial f_3}{\partial x} & \frac{\partial f_3}{\partial y} & \frac{\partial f_3}{\partial z} & \frac{\partial f_3}{\partial D} \end{bmatrix} \tag{6}$$

A further unknown quantity is the time delay introduced by the receiving arrangement due to unknown delays, e.g. due to error recovery mechanisms.

From WO 00/48018 it is known to use two separate receiving arrangements in one transmission/receiving station to compensate this kind of unknown delays. Fig. 3 shows a corresponding transmission/receiving station with a compensation of the time delay introduced by the receiving arrangement. The transmission and receiving station 301 comprises a multiplexer/encoder 302, a QPSK modulator 303, an up-converter 304 and a satellite antenna 305. Digital payload signals 306 consist of elementary data streams and are fed to a multiplexer/encoder 302 which converts the plurality of digital payload signals into a single digital transport stream, for example according to the MPEG-2 and DVB standards. The digital transport stream is modulated by the QPSK modulator 303 and fed to the up-converter 304 which represents the equipment necessary to convert the

output of the QPSK modulator 303 into a signal that can be fed to the satellite antenna 305 for transmission to a transponder carried by the satellite. Typically, such a transponder comprises frequency converters, high-power
5 amplifiers etc.

The output signal of the QPSK modulator 303, i.e. the modulated digital transport stream is also fed to a first receiving arrangement 307. The processor 308 analyses the
10 series of samples to trace a predetermined signal pattern. If the predetermined signal pattern is traced, the processor 308 sends a start signal START to a time measurement circuit 309. Upon receipt of the start signal START the time measurement circuit 309 begins to measure
15 the time until it receives a stop signal STOP.

The stop signal STOP is generated by a second processor 308' receiving an output signal from a second receiving arrangement 307'. The first and second receiving
20 arrangements 307, 307' are identical regarding their structure and components. The input signal to the second receiving arrangement 307' is supplied from a down-converter 310 which receives a signal from the satellite antenna 305 and which comprises all the equipment necessary
25 to convert the received signal from the satellite antenna 305 into a signal corresponding to the output signal of the QPSK modulator 303.

However, since the signal has travelled from the satellite
30 antenna 305 to the transponder carried by the satellite 312 and back, the received signal is delayed.

For generating the stop signal STOP, the second processor 308' traces the predetermined bit sequence in the output
35 signal of the second receiving arrangement 307' in the same

manner as the first processor 308. Upon detection of the predetermined bit sequence, the second processor 308' sends the stop signal STOP to the time measurement circuit 309 which stops the time measurement. The measured time
5 corresponds to the double distance between the ground station 305 and the transponder carried by the satellite 312, wherein fixed time delays in the up-converter 304, the satellite antenna 305, the transponder carried by the satellite 312 and the down-converter 310 can be subtracted
10 accordingly. Since two identical receiving arrangements 307, 307' are provided, unknown delays, e.g. due to error recovery mechanisms, can be compensated accordingly.

Fig. 4 shows a diagram of a transport stream according to
15 the MPEG-2 standard. The transport stream TS is a sequence of packets basically consisting of a header H (4 bits) and a payload P (184 bits). The header H includes synchronisation information (1 bit), various flags (transport error indicator, payload unit start indicator,
20 transport priority, etc.), a payload identification PID (13 bits) and a continuity counter (4 bits). The payload identification PID is required for demultiplexing the individual elementary data streams. An adaptation field is optional, but is transmitted at least every 0.1 s and
25 contains ancillary program data, especially a program reference clock PCR for regeneration of a 27 MHz clock at the receiving side.

Subsequently, the transport stream TS is processed
30 according to different standards depending on the transmission channel. For transmission via satellites, the European DVB satellite standard (DVB-S) may be applied, which defines among other mechanisms convolutional and read-solomon coding as well as additional error control
35 bits to be added to allow forward error correction (FEC).

Similarly, European DVB standards exist for terrestrial (DVB-T) and cable (DVB-C) broadcasting.

5 The predetermined bit sequence in the transport stream TS can be used to generate trigger signals or predetermined signal patterns on the basis of which the delay caused by the travel path from the satellite ground station to the transponder carried by the satellite and back can be calculated. The predetermined bit sequence may be inserted
10 into the transport stream TS at the uplink side, for example as a specific payload P. In order to avoid insertion of additional packets, the program identification PID or part of it may be used as a predetermined bit sequence. Some PIDs must be present in the transport stream
15 TS, but may have a repetition rate which is too high for the purpose of determining ranging information. Then, the PID may be combined with other information of the transport stream header H, e.g. the continuity counter, in order to define a predetermined bit sequence.

20 On the basis of the configuration according to Fig. 2, each station can carry out the distance measurement on its own. In a next step, the position of the satellite can be calculated on the basis of formulas (5) and (6) as
25 described above in a central processing station. However, in order to provide independent stations, for each station a transmission equipment has to be provided which makes considerable investments necessary.

30 An alternative solution is the use of a so-called pseudo ranging configuration. Fig. 5 shows the practical configuration of a quadrilateration measurement by pseudo ranging. Pseudo ranging in the sense of the present application is the simple delay measurement of the time
35 elapsed between the transmission of a signal from one

location and the reception of the same signal to another location. In practice, one transmission station and a plurality of receiving stations are established as shown in Fig. 5. Preferably, one receiving station is combined with the transmission station. However, in order to determine for each station the transmission delay, it is now necessary to introduce a time synchronization between all stations. Only if the receiver also knows when the transmitter actually has sent the signal, it is possible to measure the delay or so-called pseudo range. In the case of geo-stationary satellite ranging, the delay is still relatively short (about 250 ms) , so that the clock synchronization needs to be highly stable for short periods. More specifically, the synchronization accuracy should be at least below 10 ns.

Naturally, the same compensation technique as known from WO 00/48018 can be applied also for pseudo ranging. Fig. 6 shows separate transmission and receiving stations with a compensation of the time delay introduced by the receiving arrangement. The transmission station 601 comprises a multiplexer/encoder 602, a QPSK modulator 603, an up converter 604 and a first satellite antenna 605. Digital payload signals 606 are elementary data streams and are fed to the multiplexer/encoder 602 which converts the plurality of digital payload signals into a single digital transport stream, for example according to the MPEG-2 and DVB standards as described with reference to Fig. 3. The digital transport stream is modulated by the QPSK modulator 603 and fed to the up converter 604 which represents the equipment necessary to convert the output of the QPSK modulator 603 into a signal that can be fed to the satellite antenna 605 for transmission to the transponder carried by the satellite 614. Typically, such equipment comprises frequency converters, high-power amplifiers etc.

The output signal of the QPSK modulator 603, i.e. the modulated digital transport stream is also fed to a first receiving arrangement 607. The output signal of the receiving arrangement 607 is processed by a processor 608, which traces the output signal for a predetermined signal pattern. If the processor 608 traces the predetermined signal pattern, it sends a first trigger signal EMISSION to a time measurement circuit 609. Upon receipt of the first trigger signal EMISSION, the time measurement circuit 609 registers the time stamp information (the emission time) supplied by a first clock circuit 611.

Furthermore, the receiving station 612 comprises a second satellite antenna 613 and a down-converter 610' which receives a signal from the second satellite antenna 613 and which comprises all the equipment necessary to convert the received signal from the satellite antenna 613 into a signal corresponding to the output signal of the QPSK modulator 603. However, since the signal has travelled from the first satellite antenna 605 via the transponder carried by the satellite 614 to the second satellite antenna 613, the received signal is delayed.

The output signal of the down converter 610' is supplied to a second receiving arrangement 607'. The first and second receiving arrangements 607, 607' are identical regarding their structures and components, i.e. regarding their influences on the processed signal. The second processor 608' receives the output signal of the second receiving arrangement 607' and traces the output signal for a predetermined signal pattern. Upon detection of the predetermined signal pattern, the second processor 608' sends a trigger signal RECEPTION to a time measurement circuit 609' which registers the time stamp information

supplied (i.e. the reception time) by a second clock circuit 611'.

5 The second time measurement circuit 609' then transmits the time stamp information (the reception time) to the first time measurement circuit 609 which calculates the signal delay on the basis of the time stamp information received from the second time measurement circuit 609' and the time stamp information (the emission time) previously registered
10 by the first time measurement circuit 609.

For future telecommunication satellites, it is planned to replace the currently used wide area beams by narrow spot beams.

15 Fig. 7 shows a satellite carrying a conventional transponder with a global coverage. The antenna of the transponder has a 3 dB beam width of 17.5° so that the visible part of the earth is covered by the wide area beam.
20 Therefore, all transmission stations are located under one wide area beam, wherein configurations according to Fig. 2 or Fig. 5 are possible. However, as already explained, for future satellites it is planned to replace the currently used wide area beams by narrow spot beams.

25 Fig. 8 shows a satellite carrying a transponder which is connected to an antenna having several narrow beams. By way of an example, a 3 dB beam width of 1.75° is shown. For example, the narrow spot beam 801 may be used as an uplink transmission path, whereas one or more narrow spot beams
30 802 may be used as a downlink transmission path. However, for these future satellites, which will only have narrow, asymmetric spot beams, the problem is that a ranging station could only transmit, but not receive its own
35 ranging signal, unless it is located at the intersection of

a transmit and a receive beam. This means that a configuration according to Fig. 2 is not possible, but only a configuration according to Fig. 5. Furthermore, due to the small spot beam, the spatial separation between the different pseudo ranging stations is severely reduced and thus the precision of the orbit determination. Especially for co-locating multiple satellites in a single orbit slot (satellite cluster), the accuracy will not be sufficient any more.

Fig. 9 shows the effect of reduced spatial separation between different ranging stations. A two-dimensional representation is shown for simplifying matters which can be easily extended to the three-dimensional case. On the left-hand side a configuration is shown with optimum spatial separation. Two stations (station I, station II) are located on earth at a distance given by the baseline b . Each station performs a ranging measurement, either by a two-way configuration according to Fig. 2 or by a pseudo ranging configuration according to Fig. 5. It can be shown that the uncertainty of the measurement is different with regard to the propagation direction. Across the propagation direction there is a relatively low uncertainty of the ranging measurement, whereas along the propagation direction there is a relatively high uncertainty of the ranging measurement. This effect is symbolised by two error ellipses, one error ellipse for station I and one error ellipse for station II. When the two beams of the stations I and II intersect at right angle, the error in the target position of the satellite may be described as a circle which is given by the intersection of both error ellipses. This means that the high uncertainty along the propagation direction is compensated by the low uncertainty across the propagation direction of the other station.

On the right-hand side of Fig. 9, a configuration is shown with a low spatial separation between two different ranging stations I and II. In this case, the angle of intersection is much below a right angle so that the high uncertainty
5 along the propagation direction cannot be compensated anymore by the other station. Therefore, the intersecting error at the target position of the satellite can be described by a common error ellipse of the stations I and II. This results in a high uncertainty of the ranging
10 measurement across the propagation direction of both stations I and II.

Therefore, it is an object of the invention to provide a ranging system for determining ranging information of a
15 satellite carrying a transponder as well as to provide a method thereof which yield a sufficient accuracy without causing further costs when narrow spot beams by the transponder are used.

20 This object is solved by a ranging system and a ranging method according to the claims 1 and 2. Further solutions are given by a receiving station and a method for operating a receiving station according to claims 3 and 4, a processing station and a processing method according to
25 claims 5 and 6, a spacecraft and a method for operating a spacecraft according to claims 7 and 8, and a transmission unit and a method for operating a transmission unit according to claims 9 and 10.

30 According to the invention a ranging system is based on a pseudo ranging configuration as shown in Fig. 5 comprising a synchronisation means for providing a synchronised time base at least between the plurality of receiving stations. Furthermore, in addition to the synchronised time base at
35 least one receiving station comprises a correlation

receiver for receiving the reference signal. The at least one receiving station comprising the correlation receiver can be positioned outside the coverage area of the payload signal. Thereby, the base line between different ranging stations can be increased which yields a larger angle of intersection and thus a higher spatial separation as explained according to Fig. 9.

Fig. 10 shows a satellite carrying a transponder which yields a coverage 1001 by a spot beam. However, a receiver for receiving a reference signal can be placed outside the coverage 1001 by using a correlation receiver according to the invention. In principle, such a receiver can be placed inside the coverage 1002. The spot beam 1001 may have again a 3 dB beam width of 1.75° and may be used for transmitting both a payload signal and the reference signal. In order to enable the placement of several receiving stations outside the footprint of the spot beam 1001, receiving stations 1003 and 1004 are provided with correlation receivers for receiving the reference signal. It has to be noted that the processing of the reference signal is not time crucial and has not to be performed in real time in contrast to the payload signal. Thereby, a long predetermined signal pattern can be provided which can be processed offline. The higher the correlation gain is, the lower the signal-to-noise ratio of the received signal may be, which means in other terms, the further away the receive station may be from the footprint 1001.

Fig. 11 shows the antenna pattern of a satellite antenna. The effective isotropic radiated power (EIRP) is plotted versus the azimuth angle (AZ) or elevation angle (EL), respectively. The chart can also be interpreted as the change of EIRP depending on the position of a corresponding receiving station with regard to the centre of the

footprint of the satellite antenna. For example, if a receive station is moved out of the main beam by 0.75 deg (which corresponds to 500 km at a distance of 38400 km), the receive power will drop from 50 to 46.7 dBW for the payload signal. Moving further out to about 1.25 deg (about 840 km) from the footprint centre, the EIRP will drop to 40 dBW.

However, not the EIRP but the carrier to noise ratio (C/N) is the essential value for deciding upon the quality of the received payload signal. Therefore, by way of an example, Fig. 12 shows the relationship between the downlink EIRP and the C/N ratio of a signal received at a receiving station. Curve A shows the relationship for a usual payload signal of 36 MHz bandwidth and curve B shows the relationship for a spread spectrum signal with very high processing gain which is transmitted simultaneously with a payload signal. For both curves it is assumed, that the payload signal is received via a 1.2 m dish from a Ku band transponder.

In a first step it is obvious, that the carrier to noise ratio C/N of the payload signal versa the EIRP follows a linear relationship according to curve A. It can be assumed, that a minimum C/N of 6 dB is needed to receive a payload signal properly. Hence, a corresponding EIRP of 40 dBW is required at the receiving station. From the explanations according to Fig. 11 follows, that a receiving station cannot be positioned further than 840 km from the centre of the satellite footprint assuming a geo-stationary satellite at 38400 km to receive a payload signal with a 1.2 m dish. Consequently, if one wants to increase the distance to the centre of the footprint due to accuracy reasons as explained according to Fig. 9, it is necessary to increase the antenna gain and thus the diameter of the

antenna dish. For example, if one wants to position a receiving station at the first side lobe according to Fig. 11, about 20 dB of loss in EIRP have to be compensated by an increased antenna gain. This results in a necessary dish size of about 10 m having 61 dB antenna gain compared to the 1.2 m antenna dish having 41 dB antenna gain for Ku-band.

However, according to the invention a correlation receiver is used which yields a "correlation gain" and therefore allows the processing of a reference signal without increasing the antenna size, even if the payload signal has a negative C/N. All kinds of correlation methods which yield a correlation gain are covered by the invention. Two correlation methods will be discussed in more detail according to Fig. 12, namely the processing of a predetermined signal pattern and the spread spectrum modulation and demodulation.

The use of a predetermined signal pattern which is introduced in the payload signal or in a separate reference signal leads in principle to the same curve A as explained above. However, the curve A is now shifted upwards due to the introduced correlation gain. Therefore, even a negative C/N according to curve A can now be compensated by a corresponding correlation gain, hence enabling to use the same antenna size. It has to be noted that the processing time of the reference signal has not to be done in quasi-real time as it is necessary for the payload signal. Therefore, the receiving stations can well be located outside a given footprint of a narrow spot beam, if the correlation receiver accepts that its signal is received with a very low or even negative signal-to-noise ratio. A processing is still possible, if the time of reception of the predetermined signal pattern is known and is correlated

with the predetermined signal pattern in the correlation receiver. In this case, the length of the ranging signal and the height of the correlation peak between transmit and receive signal determines the acceptable level of noise on the receive signal. The higher the correlation gain, the lower the signal-to-noise ratio of the received signal may be, which means in other terms the further away the receive station may be from the main footprint. In this way, the distance between the different receiving stations may be considerably increased.

Another correlation technique is the spread spectrum modulation and demodulation. In this case the payload signal may be a modulated carrier signal, whereas the reference signal is a spread spectrum signal. In spread spectrum modulation and demodulation, the transmitted modulation is spread (increased) in bandwidth prior to transmission over the channel and then despread (decreased) in bandwidth by the same amount at the receiver. By far the most popular spreading techniques are direct sequence (DS) modulation and frequency hopping (FH) modulation.

A direct sequence modulation is formed by linearly modulating the output sequence of a pseudo random number generator onto a chain of pulses, each having a duration called the chip time. This type of modulation is usually used with binary phase-shift-keyed (BPSK) information signals. As such the modulated signal is formed by first multiplying (modulo-2) the pure information bit stream with the pseudo noise sequence and then modulating the phase of a clean carrier signal with the resulting signal.

At the receiver, the PN waveform usually is available at the receiver (however, there are also application which acquire the PN waveform from the received signal) .The PN

waveform at the receiver is used to despread the received signal again to its original bandwidth.

A frequency hopping modulation is formed by non-linearly modulating a chain of pulses with a sequence of pseudo randomly generated frequency shifts. This modulation signal is multiplied by a complex multiple-frequency-shift-keyed (MFSK) information signal. At the receiver, the sum of the transmitted signal and the channel interference is complex multiplied by the identical frequency hopping modulation which returns the transmitted signal to its original MFSK form. Analogous to the direct sequence case, the receiver knows the dehopping waveform or must alternatively acquire and track the frequency-hopped signal.

The use of a spread spectrum modulation of the reference signal leads to curve B in Fig. 12. Curve B differs from curve A, because the reference signal RS is not only disturbed by the noise N as seen from the antenna, but also by the payload signal itself. For high EIRP values, the degradation for the ranging signal is dominated by the carrier level due to $C \gg N$. For instance, at high EIRP the C/N is about 12 ... 14 dB typically. If the receiving station is removed out of the footprint, C and RS are attenuated by the same amount. On the other hand, N remains constant. This explains, why $RS/(C+N)$ decreases far less than C which makes it possible to use spread spectrum modulation also as a correlation technique. For example, in the footprint centre a C/N of about 16 dB and a $RS/(C+N)$ of -25dB is obtained. If the processing gain of the reference signal is about 40 dB, this signal still can be received and processed. At an EIRP of 40 dBW, the C/N decreases by 10 dB to about 6 dB, whereas the spread signal merely decreases to -25.8 dB, which is only 0.8 dB less than in the centre of the footprint. Furthermore, if the EIRP drops by 20 dB

to 30 dBW, the C/N of the payload signal decreases by 20 dB to -3dB, which makes a standard video signal impossible to decode. However, the reference signal decreases only by 5 dB to about -30 dB which is still possible to decode
5 assuming a processing gain of >35 dB.

A further correlation technique for receiving an MPEG data stream is known from EP 1 041 397 A2. Usually, a
predetermined signal pattern in an MPEG stream cannot be
10 used for triggering a time measurement element in the receiver because the time resolution is not sufficiently high due to non-deterministic signal delays in the decoder. In order to increase the accuracy, a two-stage method is proposed, using a combination of a direct processing of the
15 MPEG stream after the decoder and a direct correlation of the analogue signal before the decoder. The direct processing of the MPEG stream after the decoder yields a rough estimation of the propagation time of the predetermined signal pattern including the non-
20 deterministic signal delays in the decoder. Based on this rough estimation, a high-precision correlation is performed on the digitised analogue signal before the decoder. The result of the high-precision correlation is used for correcting the rough determination. However, this method
25 still needs a sufficient C/N to enable the rough processing of the demodulated signal. If no rough processing is possible in the first stage, the exact signal pattern of the modulated signal behind the LNB and before the demodulator has to be known in order to be able to apply a
30 correlation on the analogue reference signal.

Eventually, it goes without saying that all combinations of correlation techniques are also possible in order to increase the correlation gain even further. In particular,
35 it is possible to use a spread spectrum reference signal

containing a predetermined signal pattern, wherein the despread signal is again subject to a cross-correlation with the pre-stored predetermined signal pattern or subject to an autocorrelation.

5

Hence, in all cases the use of a correlation receiver in a receiving station has the advantage that the additional correlation gain and/or processing gain may compensate for lack of signal to noise ratio, and thus leads to a
10 reception of the reference signal without increasing the antenna size.

There are several possibilities for providing a time synchronization between the different receiving stations.

15

One possibility is to provide a high-precision GPS time transfer. The global positioning system (GPS) is not only a navigation system, but it is also a time transfer system. The present accuracy of GPS time transfer is of the order 10 -20 ns for intercontinental distances and 2 -3 ns within
20 one continent. Various GPS time transfer techniques have been proposed e.g. in *GPS Time Transfer*, Wlodzimierz Lewandowski and Claudine Thomas, *Proceedings of the IEEE*, vol. 79, No. 7, July 1991. Another technique of time transfer is the so-called two-way time transfer via
25 communication satellites, which is further explained e.g. in *Two-Way Time Transfer via Communication Satellites*, Dieter Kirchner, *Proceedings of the IEEE*, vol. 79, NO. 7, July 1991. In contrast to the usual one-way mode of operation, in the two-way technique the stations involved
30 in a time transfer have to exchange timing signals via satellite and therefore require receive- and transmit stations, e.g. like a configuration according to Fig. 2. In general this means more expensive equipment and more elaborate operational procedures than with the one-way
35 method. Moreover, because the two-way mode is a point-to-

point technique, the users have to work in pairs and need to exchange their measurement data. The main advantage of the two-way time transfer technique is that in contrast to the one-way method the knowledge of the satellite and user
5 positions is not the limiting factor for the obtainable accuracy, because for reciprocal, paths between the stations the path delays cancel out and have therefore not to be calculated. Because of this fact, the two-way method has the potential to be the most accurate time transfer
10 method and hence there might be some applications even for the two-way time transfer method.

According to a further aspect of the present invention, a time synchronization method is proposed which is not known
15 in prior art. This new time synchronization method is based on the cognition that a time synchronization is provided by a common time base in combination with a correction of the synchronization error of the common time base by a reference ranging information of a reference satellite.

20 Hence, besides the satellite whose position has to be determined, another reference satellite is needed whose position is known precisely in advance. If such a reference satellite is available, it can be observed that the synchronization error for one receiving station with regard
25 to the central transmission station is approximately the same for the two satellite. This means that even a common time base with a usually non-sufficient time synchronization can be used by compensating the time synchronization errors using the ranging measurement of the
30 reference satellite whose position is known.

According to the present invention, there are several possibilities for locating the receiving stations with regard to the footprint centre of the transponder antenna
35 carried by the satellite. In any case it will be an aim to

locate the receiving stations such to achieve an optimal accuracy of calculation of said ranging information. The usual case might be a configuration of one transmission station and four receiving stations, wherein the
5 transmission station comprises already one receiving station. For this configuration an optimal accuracy of calculation can be achieved, if three receiving stations are located equispaced on a circle and the transmission station in the middle, thereby forming an equilateral
10 triangle with regard to the satellite.

It is understood that the present invention is not limited to a satellite carrying a transponder, but that the ranging of a communications satellite is only a preferred aspect of
15 the present invention. In general, the coordinates of any spacecraft can be determined according to the invention, which carries a component of a communication channel which allows the measurement of a transmission delay between the component and a plurality of receiving stations. This
20 means, that such a component can be a transponder, but also a repeater, mirror, etc. Furthermore, the term "spacecraft" in the sense of the present invention covers any moving or movable object.

25 According to a further solution of the invention a control unit and a method for operating a control unit is provided for controlling the position of a spacecraft carrying a component of a communication channel. The functionality of a control station for a satellite is summarized by the
30 terms telemetry, tracking and command (TTC). TTC deals with the following:

- Receiving control signals from the ground to initiate manoeuvres and to change the state or mode of
35 operation of equipment.

- Transmitting results of measurements, information concerning satellite operation, the operation of equipment and verification of the execution of commands to the ground.
- Enabling measurement of the ground-satellite distance, and possibly the radial velocity, in order to permit location of the satellite.

In summary, the above-identified object of the invention is solved by the following features, which are listed in dependency order according to the subject subject matters mentioned above:

1. Ranging system for determining ranging information of a spacecraft carrying a component of a communication channel, comprising:

a plurality of receiving stations at different locations on earth, wherein each receiving station is arranged for receiving a reference signal from said component;

synchronisation means for providing a synchronised time base between the plurality of receiving stations;

calculation means for calculating said ranging information in accordance with the propagation time of each received reference signal and with the synchronised time base;

wherein at least one receiving station comprises a correlation receiver for receiving the reference signal.

2. Ranging system according to subject matter 1, wherein
the component is a transmitter transmitting the
reference signal to the plurality of receiving
stations.
3. Ranging system according to one of the subject matters
1 - 2, wherein the component is a transponder and
wherein at least one transmission station transmits a
payload signal and a reference signal via said
transponder.

Hence, any component is possible which allows an
interconnection between the spacecraft and the plurality of
receiving stations. This means, that both transmitters and
transponders can be used. Having a satellite, the
transmitter is used when no uplink path is present, whereas
the transponder is used to interconnect between an uplink
path and a downlink path.

4. Ranging system according one of the subject matters 1
- 3, wherein the antenna size of the at least one
receiving station comprising a correlation receiver is
minimised by utilising the receiving gain of the
correlation receiver.

This means, that the invention also enables the use of
receiving stations with low cost receiving components.
Futhermore, as mentioned above, the correlation receiver
allows to place the corresponding receiving station outside
the footprint of said component of the satellite for
achieving an improved accuracy of calculation of said
ranging information.

All kinds of correlation receiver are possible according to the invention, which yield a kind of correlation gain for improving the signal to noise ratio of the reference signal. In particular, the following kinds of correlation receivers are possible:

5 5. Ranging system according to one of the subject matters 1 - 4, wherein a predetermined signal pattern is stored in the correlation receiver for detecting a
10 corresponding predetermined signal pattern contained in the reference signal.

 6. Ranging system according to subject matter 5, wherein
15 the predetermined signal pattern is stored in a modulated form for detecting a corresponding predetermined signal pattern in the modulated reference signal.

 7. Ranging system according to subject matter 5, wherein
20 the predetermined signal pattern is stored in a demodulated form for detecting a corresponding predetermined signal pattern in the demodulated reference signal.

25 8. Ranging system according to one of the subject matters 1 - 7, wherein the correlation receiver is a spread spectrum receiver and the reference signal is a spread spectrum signal.

30 Another important aspect of the invention is the time synchronisation of the plurality of receiving stations. In the case of geo-stationary satellite ranging, the delay is still relatively short (about 250 ms) , so that the clock synchronization needs to be highly stable for short
35 periods. More specifically, the synchronization accuracy

should be at least below 10 ns. Preferred techniques of time synchronization are:

5 9. Ranging system according to one of the subject matters
 1 - 8, wherein the synchronisation means provides a
 high precision GPS time transfer.

10 10. Ranging system according to one of the subject matters
 1 - 8, wherein the synchronisation means provides a
 two way time transfer.

15 11. Ranging system according to one of the subject matters
 1 - 8, wherein the synchronisation means provides a
 common time base and wherein the synchronization error
 of the common time base is corrected by a reference
 ranging information of a reference spacecraft.

20 It goes without saying that the payload signal and the
 reference signal can be part of one information stream
 transmitted over the communication channel:

25 12. Ranging system according to one of the subject matters
 1 - 11, wherein the payload signal and the reference
 signal are part of one or more digital transport
 streams.

30 The subject matters 2 - 12 as described for the ranging
 system apply also for other categories of subject matters.
 Further categories of subject matters are:

 13. Ranging method for determining ranging information of
 a spacecraft carrying a component of a communication
 channel, comprising the steps of:

receiving a reference signal from said component independently by a plurality of receiving stations at different locations on earth;

5 providing a synchronised time base between the plurality of receiving stations;

calculating said ranging information in accordance with the propagation time of each received reference
10 signal and with the synchronised time base;

wherein at least one receiving station carries out a correlation operation for receiving the reference
15 signal.

14. Ranging method according to subject matter 13, wherein the component transmits the reference signal to the plurality of receiving stations.

20 15. Ranging method according to one of the subject matters 13 - 14, wherein the component repeats the reference signal to the plurality of receiving stations.

25 16. Ranging method according to one of the subject matters 13 - 15, wherein the antenna size of the at least one receiving station comprising a correlation receiver is minimised by utilising the receiving gain of the correlation receiver.

30 17. Ranging method according to one of the subject matters 13 - 16, wherein a predetermined signal pattern is stored in the correlation receiver for detecting a corresponding predetermined signal pattern contained in the reference signal.

18. Ranging method according to subject matter 17, wherein
the predetermined signal pattern is stored in a
modulated form for detecting a corresponding
predetermined signal pattern in the modulated
reference signal.
5
19. Ranging method according to subject matter 17, wherein
the predetermined signal pattern is stored in a
demodulated form for detecting a corresponding
predetermined signal pattern in the demodulated
reference signal.
10
20. Ranging method according to one of the subject matters
13 - 19, wherein the correlation operation is based on
a spread spectrum technique and the reference signal
is a spread spectrum signal.
15
21. Ranging method according to one of the subject matters
13 - 20, wherein the time synchronisation is provided
by high precision GPS time transfer.
20
22. Ranging method according to one of the subject matters
13 - 20, wherein the time synchronisation is provided
by two way time transfer.
25
23. Ranging method according to one of the subject matters
13 - 20, wherein the time synchronisation is provided
by a common time base and wherein the synchronization
error of the common time base is corrected by a
reference ranging information of a reference
spacecraft.
30
24. Ranging method according to one of the subject matters
13 - 23, wherein the payload signal and the reference

signal are part of one or more digital transport streams.

5 25. Receiving station for providing ranging information of a spacecraft carrying a component of a communication channel, comprising:

10 a correlation receiver for receiving a reference signal from said component; and

10 synchronisation means for providing a synchronised time base between said receiving station and a reference point on earth.

15 26. Receiving station according to subject matter 25, wherein the antenna size of said receiving station is minimised by utilising the receiving gain of the correlation receiver.

20 27. Receiving station according to one of the subject matters 25 - 26, wherein a predetermined signal pattern is stored in the correlation receiver for detecting a corresponding predetermined signal pattern contained in the reference signal.

25 28. Receiving station according to subject matter 27, wherein the predetermined signal pattern is stored in a modulated form for detecting a corresponding predetermined signal pattern in the modulated reference signal.

30

29. Receiving station according to subject matter 27, wherein the predetermined signal pattern is stored in a demodulated form for detecting a corresponding

predetermined signal pattern in the demodulated reference signal.

- 5 30. Receiving station according to one of the subject matters 25 - 29, wherein the correlation receiver is a spread spectrum receiver and the reference signal is a spread spectrum signal.
- 10 31. Receiving station according to one of the subject matters 25 - 30, wherein the synchronisation means provides a high precision GPS time transfer.
- 15 32. Receiving station according to one of the subject matters 25 - 30, wherein the synchronisation means provides a two way time transfer.
- 20 33. Receiving station according to one of the subject matters 25 - 30, wherein the synchronisation means provides a common time base and wherein the synchronization error of the common time base is corrected by a reference ranging information of a reference spacecraft.
- 25 34. Method for operating a receiving station for providing ranging information of a spacecraft carrying a component of a communication channel, comprising the steps of:
- 30 receiving a reference signal from said component; and
- providing a synchronised time base between said receiving station and a reference point on earth.
- 35 35. Method according to subject matter 34, wherein the antenna size of said receiving station is minimised by

utilising the receiving gain of the correlation receiver.

- 5 36. Method according to one of the subject matters 34 -
 35, wherein a predetermined signal pattern is stored
 in the correlation receiver for detecting a
 corresponding predetermined signal pattern contained
 in the reference signal.
- 10 37. Method according to subject matter 36, wherein the
 predetermined signal pattern is stored in a modulated
 form for detecting a corresponding predetermined
 signal pattern in the modulated reference signal.
- 15 38. Method according to subject matter 36, wherein the
 predetermined signal pattern is stored in a
 demodulated form for detecting a corresponding
 predetermined signal pattern in the demodulated
 reference signal.
- 20 39. Method according to one of the subject matters 34 -
 38, wherein the correlation receiver is a spread
 spectrum receiver and the reference signal is a spread
 spectrum signal.
- 25 40. Method according to one of the subject matters 34 -
 39, wherein the synchronisation means provides a high
 precision GPS time transfer.
- 30 41. Method according to one of the subject matters 34 -
 39, wherein the synchronisation means provides a two
 way time transfer.
- 35 42. Method according to one of the subject matters 34 -
 39, wherein the synchronisation means provides a

common time base and wherein the synchronization error of the common time base is corrected by a reference ranging information of a reference spacecraft.

- 5 43. Processing station for determining ranging information of a spacecraft carrying a component of a communication channel, comprising:

10 propagation time data receiving means for receiving propagation time data from a plurality of receiving stations at different locations on earth,

15 wherein said receiving stations are arranged for receiving independently a reference signal from said component, wherein a synchronised time base is provided between the plurality of receiving stations, and wherein at least one receiving station comprises a correlation receiver for receiving the reference signal.

20

44. Processing method for determining ranging information of a spacecraft carrying a component of a communication channel, comprising:

25 receiving propagation time data from a plurality of receiving stations at different locations on earth,

30 wherein said receiving stations are arranged for receiving independently a reference signal from said component, wherein a synchronised time base is provided between the plurality of receiving stations, and wherein at least one receiving station comprises a correlation receiver for receiving the reference signal.

35

45. Spacecraft, comprising:

a component of a communication channel for
transmitting or repeating a reference signal to a
5 plurality of receiving stations at different locations
on earth,

wherein a synchronised time base is provided between
the plurality of receiving stations, and wherein the
10 reference signal is arranged to be received by a
correlation receiver.

46. Method for operating a spacecraft, comprising the
steps of:

15 providing a component of a communication channel at
the spacecraft,

wherein the component is arranged for transmitting or
20 repeating a reference signal to a plurality of
receiving stations at different locations on earth and
wherein a synchronised time base is provided between
the plurality of receiving stations,

25 and wherein the reference signal is arranged to be
received by a correlation receiver.

47. Transmission unit, comprising:

30 first transmission means for transmitting a reference
signal to a plurality of receiving stations via a
transponder carried by a spacecraft, wherein the
reference signal is arranged to be received by a
correlation receiver, and

synchronisation means for providing a synchronised time base between the transmission unit and the plurality of receiving stations,

5 48. Transmission unit according to subject matter 47,
 wherein the ranging information of said spacecraft is
 calculated in accordance with the propagation time of
 the reference signal between said transmission unit
 and each of the plurality of receiving stations and
10 with the synchronised time base.

 49. Transmission unit according to subject matter 48,
 comprising a second transmission means for
 transmitting a command signal to said transponder to
15 change the state or mode of operation of said
 spacecraft depending on the ranging information.

 50. Method for operating a transmission unit, comprising
 the steps of:

20 transmitting a reference signal to a plurality of
 receiving stations via a transponder carried by a
 spacecraft, wherein the reference signal is arranged
 to be received by a correlation receiver, and

25 providing a synchronised time base between the
 transmission unit and the plurality of receiving
 stations,

30 51. Method according to subject matter 50, wherein ranging
 information of said spacecraft is calculated in
 accordance with the propagation time of the reference
 signal between said transmission unit and each of the
 plurality of receiving stations and with the
35 synchronised time base.

52. Transmission unit according to subject matter 51, a
command signal is transmitted to said transponder to
change the state or mode of operation of said
5 spacecraft depending on the ranging information.

It is understood that all aspects of the invention
described above can be applied not only in the described
combination but also in other combinations or alone.

10 The invention will now be described by way of an example
and with reference to the accompanying drawings in which

Fig. 1 shows the basic configuration of a trilateration
15 measurement,

Fig. 2 shows the practical configuration of a
quadrilateration measurement,

20 Fig. 3 shows a transmission and receiving station with a
compensation of the time delay introduced by the
receiving arrangement,

Fig. 4 shows a diagram of a transport stream according
25 to the MPEG-2 standard,

Fig. 5 shows the practical configuration of a
quadrilateration measurement according to pseudo
ranging,

30 Fig. 6 shows separate transmission and receiving
stations with a compensation of the time delay
introduced by the receiving arrangement,

Fig. 7 shows a satellite carrying a transponder with a global coverage,

5 Fig. 8 shows a satellite carrying a transponder with a coverage by several narrow beams,

Fig. 9 shows the effect of reduced spatial separation between different ranging stations,

10 Fig. 10 shows a satellite carrying a transponder with a coverage by a spot beam and by a correlation receiver,

15 Fig. 11 shows the antenna pattern of a satellite antenna,

Fig. 12 shows the relationship between the downlink EIRP and the C/N ratio of a signal received at a receiving station,

20 Fig. 13 shows the interconnection of a processing station with remote receiving stations,

Figs. 14 - 17 show different time transfer techniques,

25 Fig. 18 shows the measurement of the time synchronisation error for different receiving stations with regard to one satellite,

30 Fig. 19 shows the measurement of the time synchronisation error for different receiving stations with regard to two different satellites,

35 Fig. 20 shows the basic configuration of a pseudo range measurement with regard to two different satellites, and

Fig. 21 shows the processing equipment for one receiving station of the basic configuration according to Fig. 20.

5

Figures 1 - 12 have been described already in the introductory part of the description.

Fig. 13 shows the interconnection of a processing station with remote receiving stations. A pseudo ranging configuration for the receiving stations may be assumed according to Fig. 5. Hence, it is assumed that the stations 1303, 1304 and 1305 are receiving stations, wherein the station 1302 is a transmission/receiving station. The internal equipment of the station 1302 may be designed according to Fig. 3, whereas the internal equipment of the stations 1303, 1304 and 1305 may be designed according to Fig. 6. It is advantageous to integrate the processing station 1301 within the transmission/receiving station 1302, wherein telecommunication links 1306, 1307 and 1308 are provided between the receiving stations 1303, 1304 and 105 and the processing station 1301. The processing station 1301 receives the propagation time data from each receiving station in order to calculate the ranging information of the corresponding satellite. The calculation is carried out on the basis of the formulas (1) - (6) as described above.

Since at least one receiving station comprises a correlation receiver, the propagation times can be fixed independently from the footprint area so that an improved accuracy of calculation of said ranging information can be achieved. Furthermore, it is possible to introduce ambiguity and plausibility checks in the processing station on the basis of the expected propagation time data.

35

Figures 14 - 17 show different time transfer techniques on the basis of the global positioning system (GPS).

Fig. 14 shows the principle of the time dissemination method. This method is the simplest and least accurate, but has a global coverage and requires no other data than those provided by the receiver. The accuracy of access to GPS time depends on local conditions of observation, mainly on the quality of the receiver antenna co-ordinates and on the amount of the required data. With antenna coordinates having an uncertainty of 10 m this accuracy is in the range of 100 ns for a 13 min track.

Fig. 15 shows the method of clock transportation. Clock transportation is of interest for high-accuracy time comparisons with 10 ns or less. The clocks A and B, at different locations anywhere on the earth, can be compared by making successive observations of the same satellite, or of the group of satellites, with delayed view times of less than 12 h. This gives the advantage of observing satellites at their maximum of elevation and so decreasing the sensitivity to models of refraction. This method is however subject to the satellite clock instabilities, which are about 5 ns over 12 h.

Fig. 16 shows the GPS common view method. Stations A and B or more stations receive the signals of the same satellite at the same time and communicate the data to each other to compare their clocks. The main advantage of this method is that satellite clock error contributes nothing because the GPS time disappears in the difference. The ultimate accuracy of common view mode is about 1 ns.

Fig. 17 shows the VLBI techniques over short base lines. This method uses the techniques of very long base

interferometry (VLBI) and can be applied for time comparisons with an accuracy below 1 ns over base lines of the order of 100 km.

5 Besides the GPS time transfer methods, another time synchronization method according to another aspect of the invention will be described according to the figures 18 - 21. The principle idea of this time synchronization method is to provide a common time base for the different
10 receiving stations, wherein the synchronization error of the common time base is corrected by a reference ranging information of a reference satellite, whose space coordinates are exactly known.

15 Fig. 18 shows the measurement of the time synchronisation error for different receiving stations with regard to one satellite. Again, the pseudo ranging configuration according to Fig. 5 is assumed, wherein at one orbital position a satellite cluster with satellites 1A, 1B, 1C,
20 1D, 1E, 1F, 1G, and 1H is provided. As can be seen from the measurement result according to Fig. 18, the time synchronisation error is not correlated between the stations III and II. However, according to the cognition of the invention, the time synchronisation error for one
25 station with regard to different satellites is nearly the same. This result is shown in Fig. 19.

Fig. 19 shows the measurement of the time synchronisation error for different receiving stations with regard to two
30 different satellites. Again referring to Fig. 5, the satellites 1A and 1B within the same orbit position are chosen, wherein the ranging error is measured over time for station III. As it can be seen on the left-hand side, the ranging errors are nearly the same, wherein at the right-

hand side the difference between both measurements is shown.

5 The result of Fig. 19 is the following: If one knows the position of one of the two satellites, let say the satellite 1B, it is possible to derive the synchronisation error between each receiving station and the transmission station. Knowing this synchronisation error, the measured pseudo range data for satellite 1A can be corrected. The
10 resulting arrangement for such a measurement is shown in Fig. 20.

Fig. 20 shows the basic configuration of a pseudo range measurement with regard to two different satellites. It is
15 assumed that station I is the master station with a transmission and a receiving equipment.

On the other hand, the other stations are receiving stations, wherein merely station III is shown by way of an
20 example. It is assumed that the position of the satellite 1A is known at the location of station I. The pseudo range measurement by the receiving station III is then performed by the following steps:

25 In a first step, two predetermined signal patterns are transmitted from the station I via the satellites 1A and 2A to the receiving station III. In a next step, the receiving station III transmits the propagation time data on the basis of a pseudo range measurement of orbit 1 and orbit 2
30 to the station I. Since according to the assumption the range of orbit 1 at the station I is known, a correction calculation can be carried out as explained according to Fig. 19. Thereby, the ranging information of the satellite 2A can be calculated and can be corrected by known
35 synchronization error with reference to the satellite 1A.

As a result, a pseudo ranging measurement can be provided on the basis of an arbitrary common time base with limited accuracy, wherein the common time base is corrected by the reference ranging information of the reference satellite
5 1A. Thereby, a GPS-independent pseudo ranging system can be built up.

Fig. 21 shows the processing equipment for one receiving station of the basic configuration according to Fig. 18. As
10 it can be seen, the receiving equipment can be built up very simply which leads to an inexpensive structure of the receiving stations. Each receiving station comprises two similar receiving arrangements comprising an LNB 2101, 2104 connected to receiving antennas 2107, 2108, a down-
15 converter 2102, 2105 and a demodulator 2103, 2106. The components are controlled by a frequency and timing device 2109, which comprises also a common data base, e.g. a simple GPS receiver. The data are collected in a data collection unit 2110 for determining the propagation time
20 data of orbit 1 and orbit 2.

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Claims

1. Ranging system for determining ranging information of
5 a spacecraft carrying a component of a communication
channel, comprising:

a plurality of receiving stations at different
locations on earth, wherein each receiving station is
10 arranged for receiving a reference signal from said
component;

synchronisation means for providing a synchronised
time base between the plurality of receiving stations;

15 calculation means for calculating said ranging
information in accordance with the propagation time of
each received reference signal and with the
synchronised time base;

20 wherein at least one receiving station comprises a
correlation receiver for receiving the reference
signal.

- 25 2. Ranging method for determining ranging information of
a spacecraft carrying a component of a communication
channel, comprising the steps of:

receiving a reference signal from said component
30 independently by a plurality of receiving stations at
different locations on earth;

providing a synchronised time base between the
plurality of receiving stations;

calculating said ranging information in accordance with the propagation time of each received reference signal and with the synchronised time base;

5 wherein at least one receiving station carries out a correlation operation for receiving the reference signal.

10 3. Receiving station for providing ranging information of a spacecraft carrying a component of a communication channel, comprising:

 a correlation receiver for receiving a reference signal from said component; and

15 synchronisation means for providing a synchronised time base between said receiving station and a reference point on earth.

20 4. Method for operating a receiving station for providing ranging information of a spacecraft carrying a component of a communication channel, comprising the steps of:

25 receiving a reference signal from said component; and

 providing a synchronised time base between said receiving station and a reference point on earth.

30 5. Processing station for determining ranging information of a spacecraft carrying a component of a communication channel, comprising:

propagation time data receiving means for receiving propagation time data from a plurality of receiving stations at different locations on earth,

5 wherein said receiving stations are arranged for receiving independently a reference signal from said component, wherein a synchronised time base is provided between the plurality of receiving stations, and wherein at least one receiving station comprises a
10 correlation receiver for receiving the reference signal.

6. Processing method for determining ranging information of a spacecraft carrying a component of a
15 communication channel, comprising:

receiving propagation time data from a plurality of receiving stations at different locations on earth,

20 wherein said receiving stations are arranged for receiving independently a reference signal from said component, wherein a synchronised time base is provided between the plurality of receiving stations, and wherein at least one receiving station comprises a
25 correlation receiver for receiving the reference signal.

7. Spacecraft, comprising:

30 a component of a communication channel for transmitting or repeating a reference signal to a plurality of receiving stations at different locations on earth,

wherein a synchronised time base is provided between the plurality of receiving stations, and wherein the reference signal is arranged to be received by a correlation receiver.

5

8. Method for operating a spacecraft, comprising the steps of:

10

providing a component of a communication channel at the spacecraft,

15

wherein the component is arranged for transmitting or repeating a reference signal to a plurality of receiving stations at different locations on earth and wherein a synchronised time base is provided between the plurality of receiving stations,

20

and wherein the reference signal is arranged to be received by a correlation receiver.

9. Transmission unit, comprising:

25

first transmission means for transmitting a reference signal to a plurality of receiving stations via a transponder, wherein the reference signal is arranged to be received by a correlation receiver, and

30

synchronisation means for providing a synchronised time base between the transmission unit and the plurality of receiving stations.

10. Method for operating a transmission unit, comprising the steps of:

transmitting a reference signal to a plurality of receiving stations via a transponder, wherein the reference signal is arranged to be received by a correlation receiver, and

5

providing a synchronised time base between the transmission unit and the plurality of receiving stations.

10

Abstract

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29 MAR 1991

The invention refers to a ranging system for determining ranging information of a spacecraft carrying a component of a communication channel. In order to provide a ranging system for determining ranging information of a satellite carrying a transponder as well as to provide a method thereof which yield a sufficient accuracy without causing further costs when narrow spot beams by the transponder are used, a ranging system according to the invention comprises a plurality of receiving stations at different locations on earth, wherein each receiving station is arranged for receiving a reference signal from said component; synchronisation means for providing a synchronised time base between the plurality of receiving stations; calculation means for calculating said ranging information in accordance with the propagation time of each received reference signal and with the synchronised time base; wherein at least one receiving station comprises a correlation receiver for receiving the reference signal.

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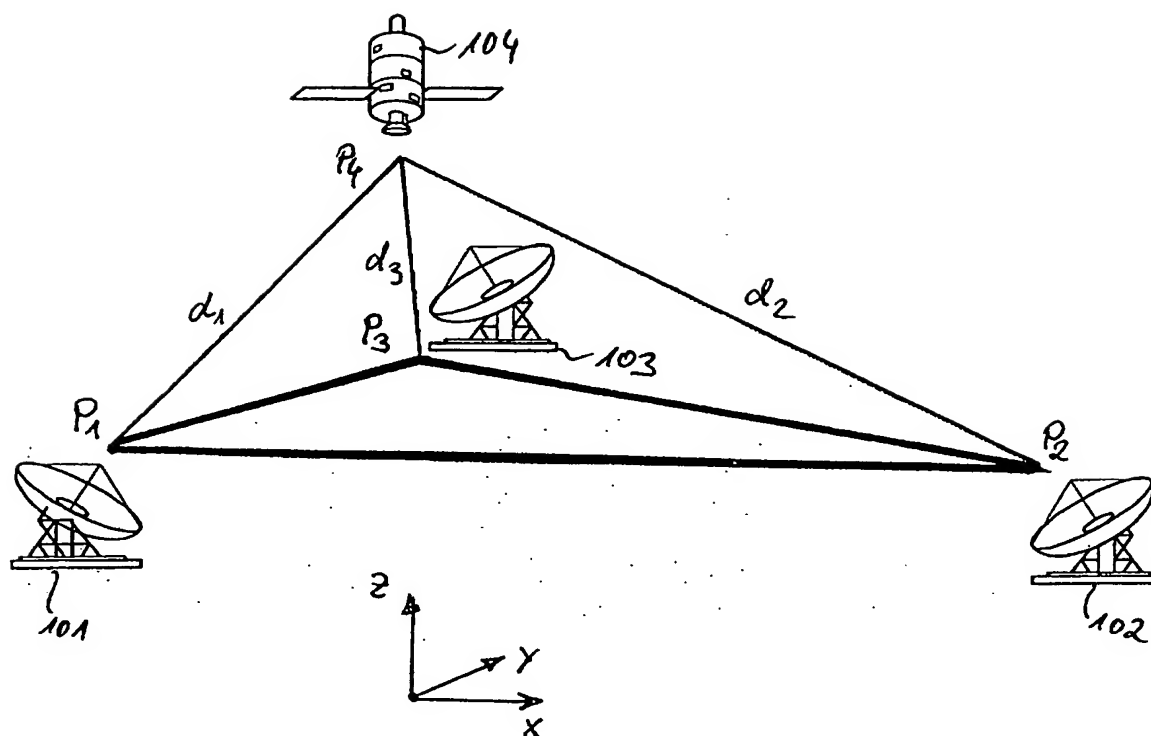


Fig. 1

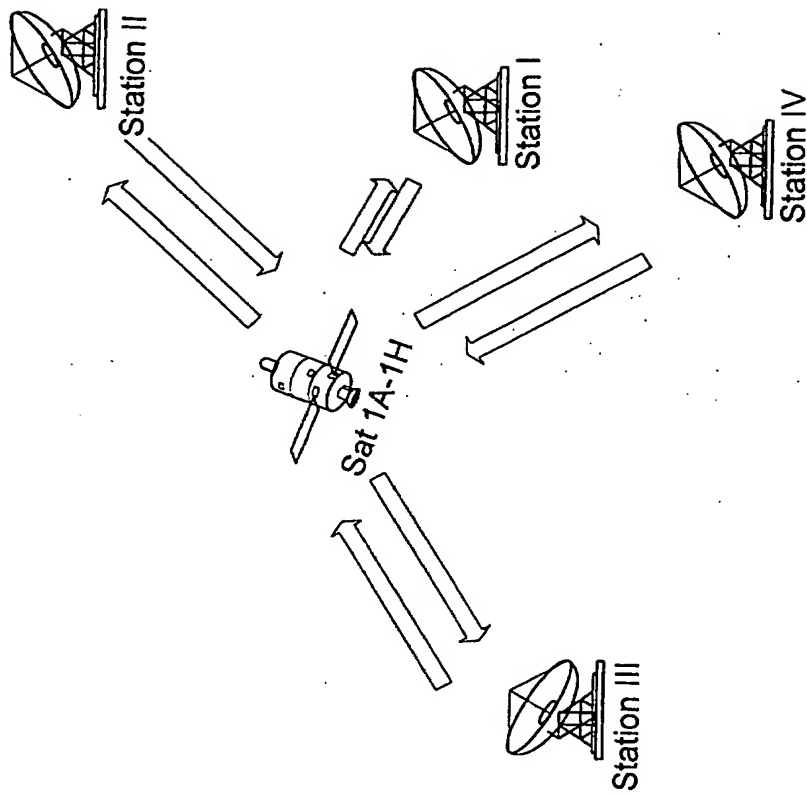
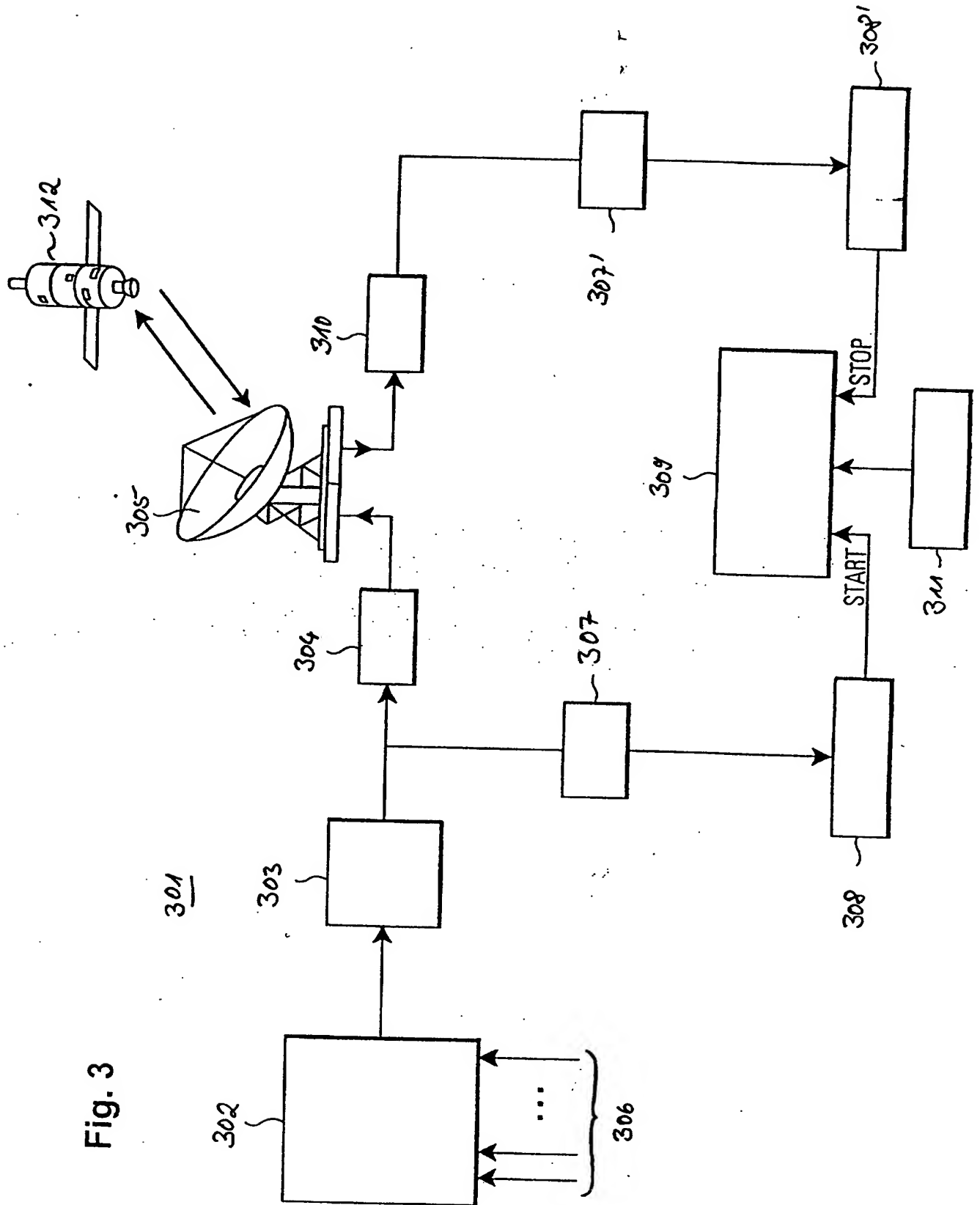
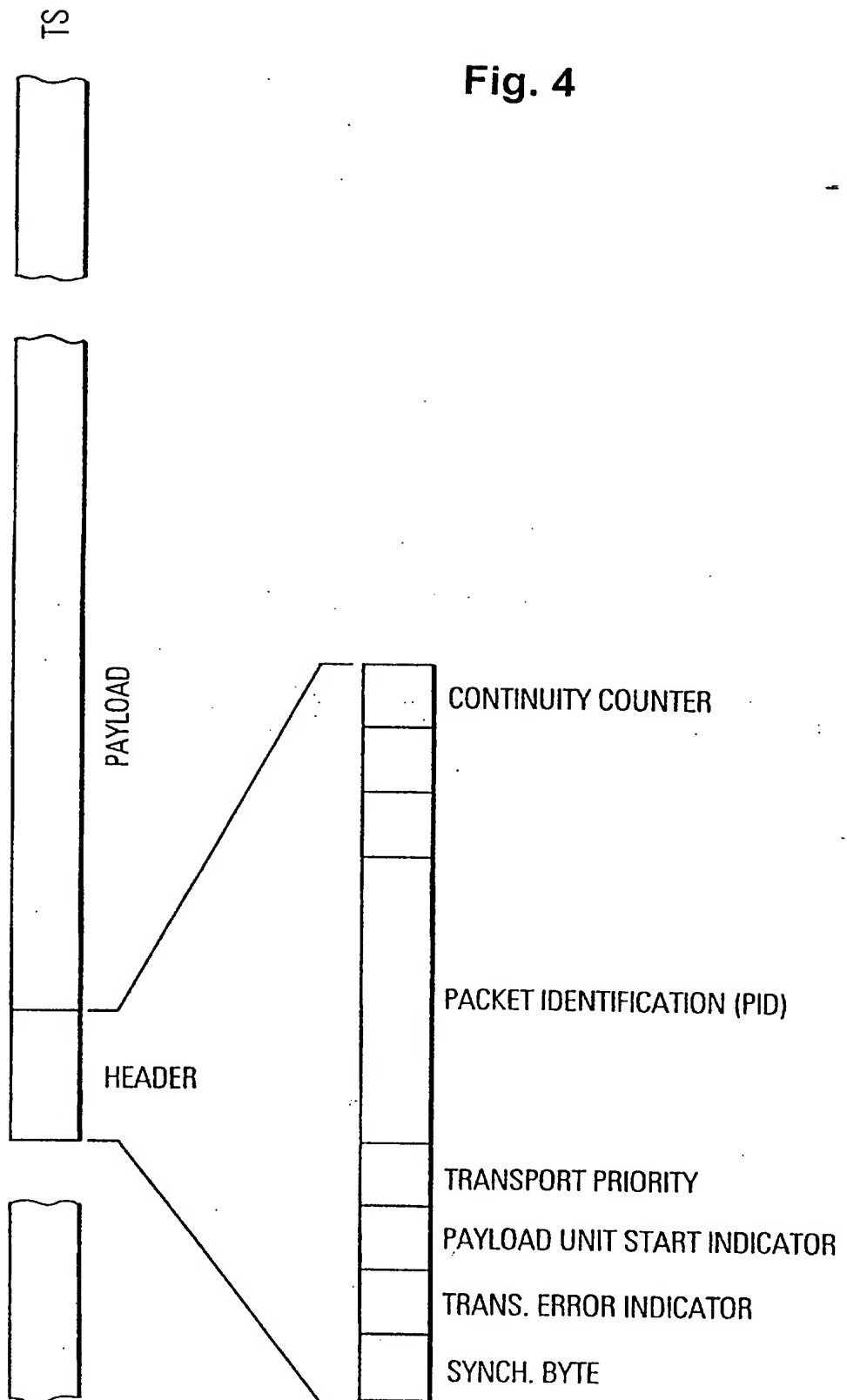


Fig. 2

Fig. 3





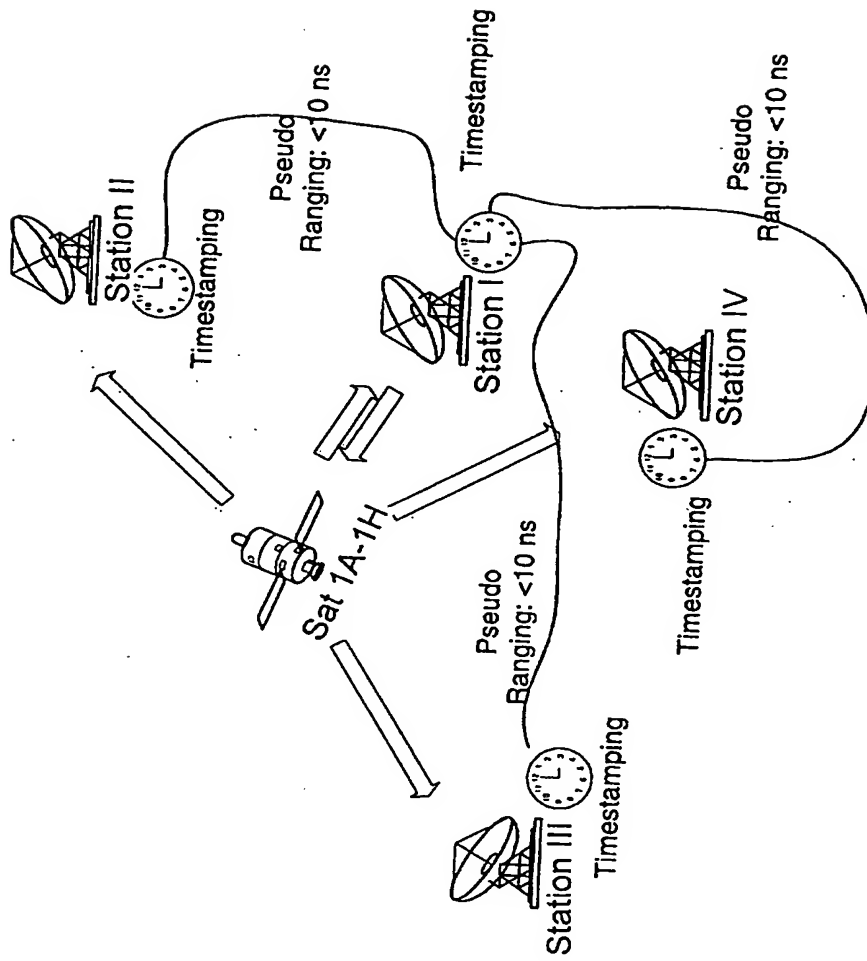
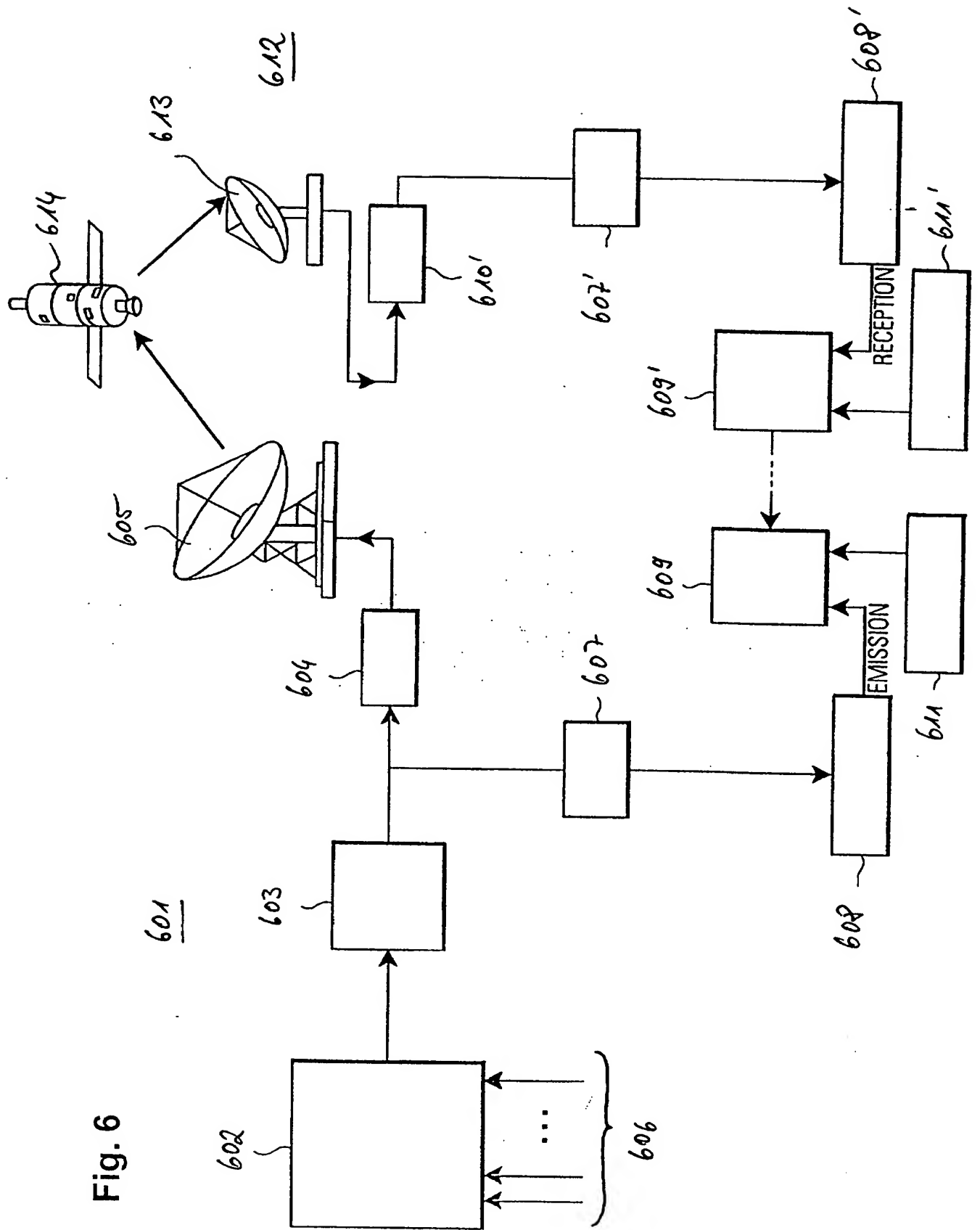
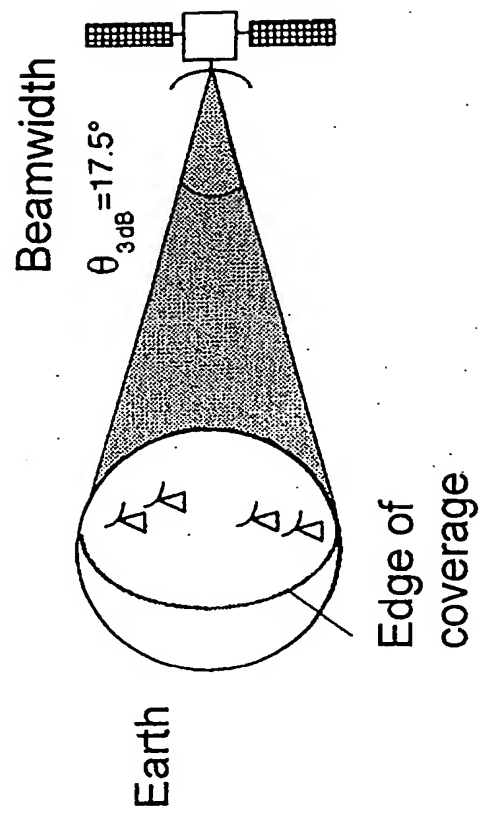


Fig. 5

Fig. 6



**Fig. 7**

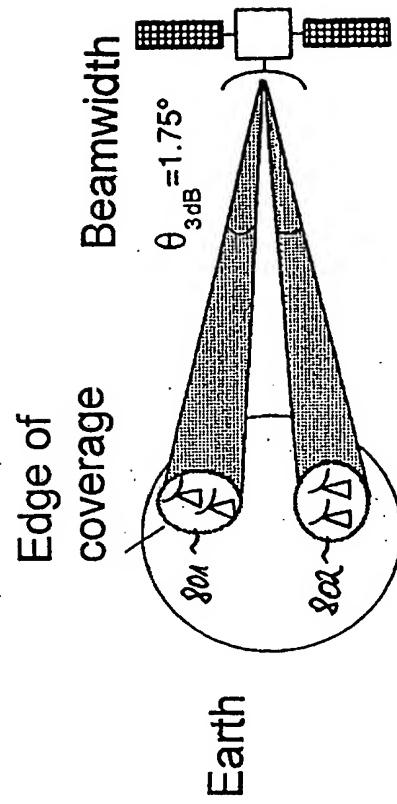


Fig. 8

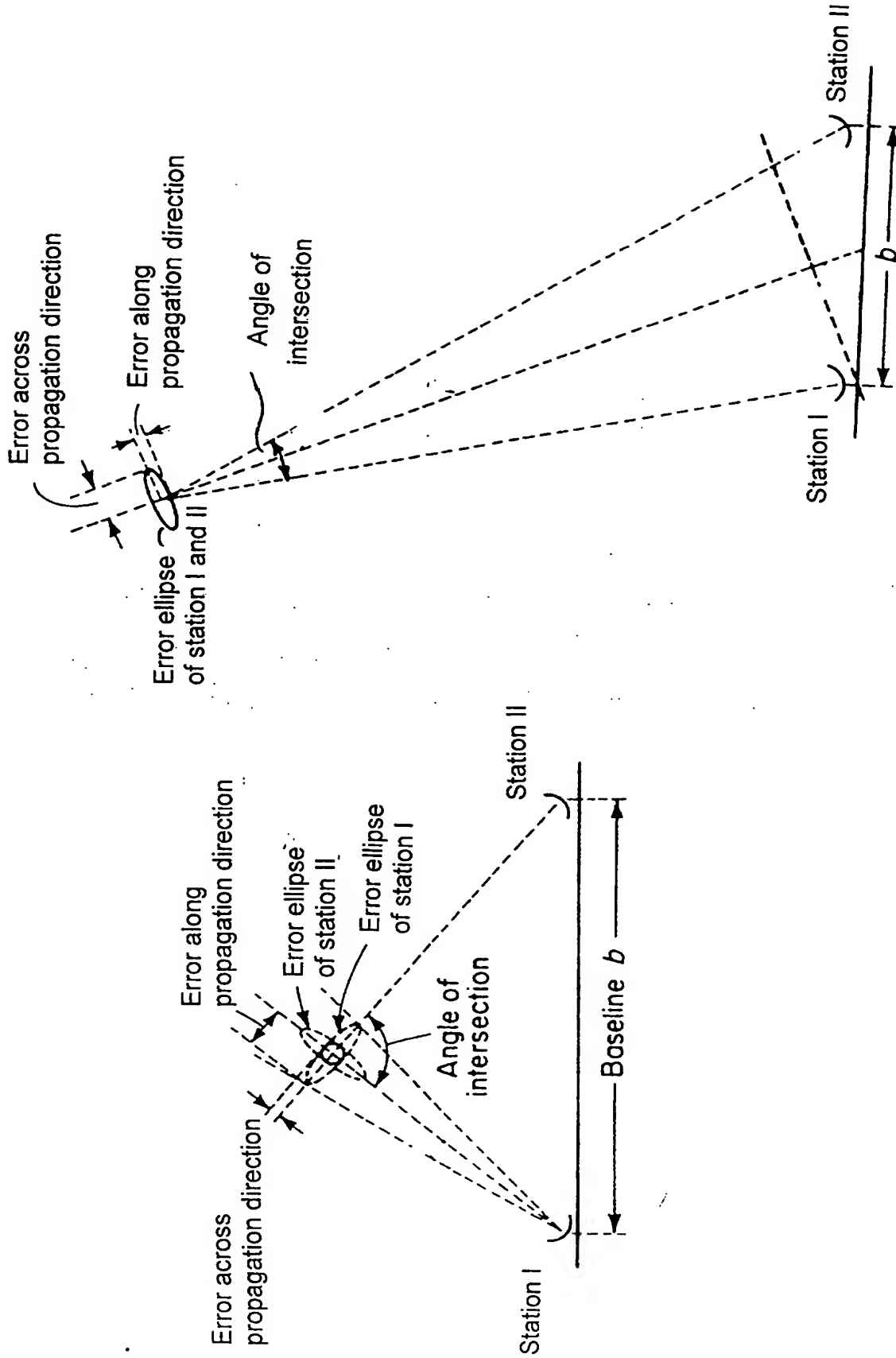


Fig. 9

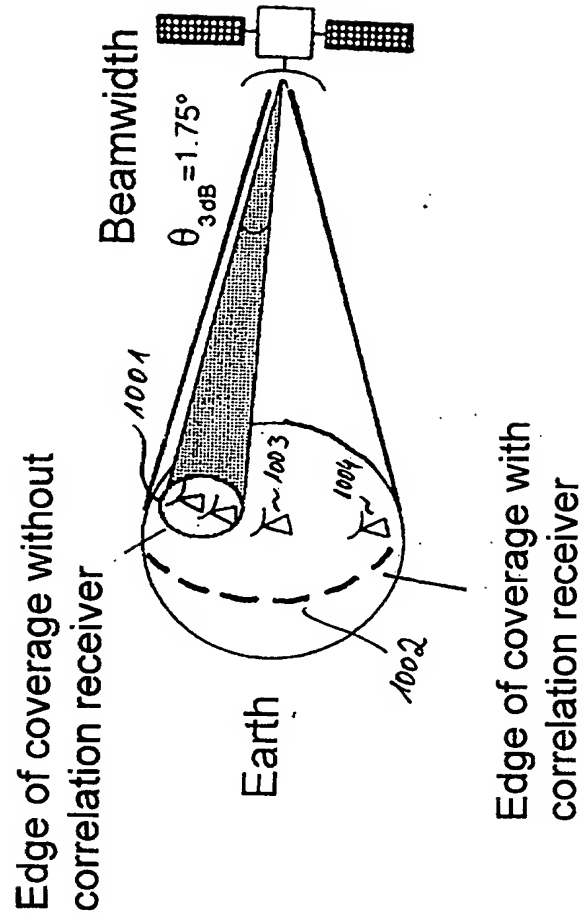


Fig. 10

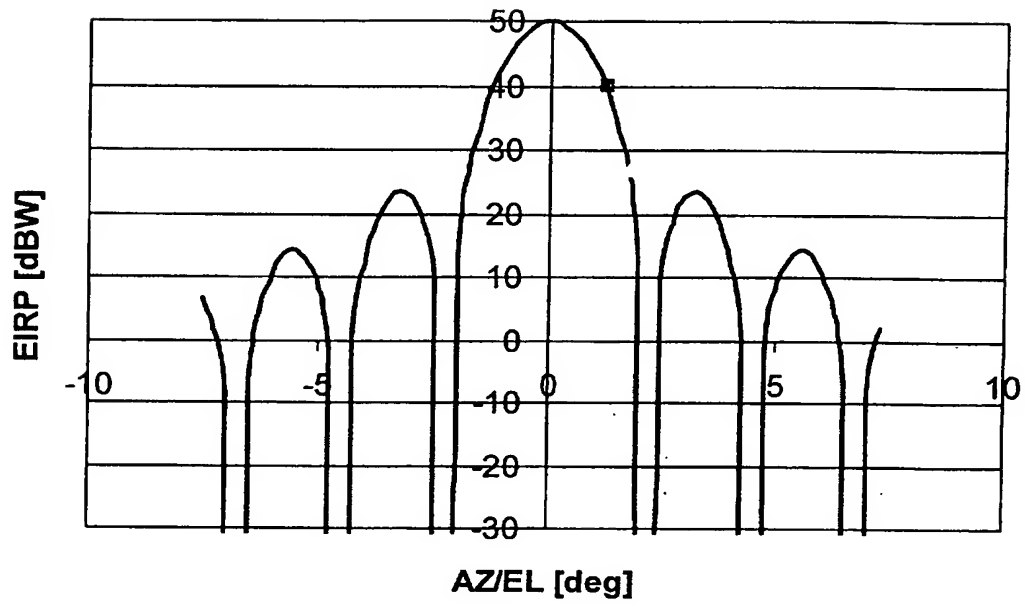


Fig. 11

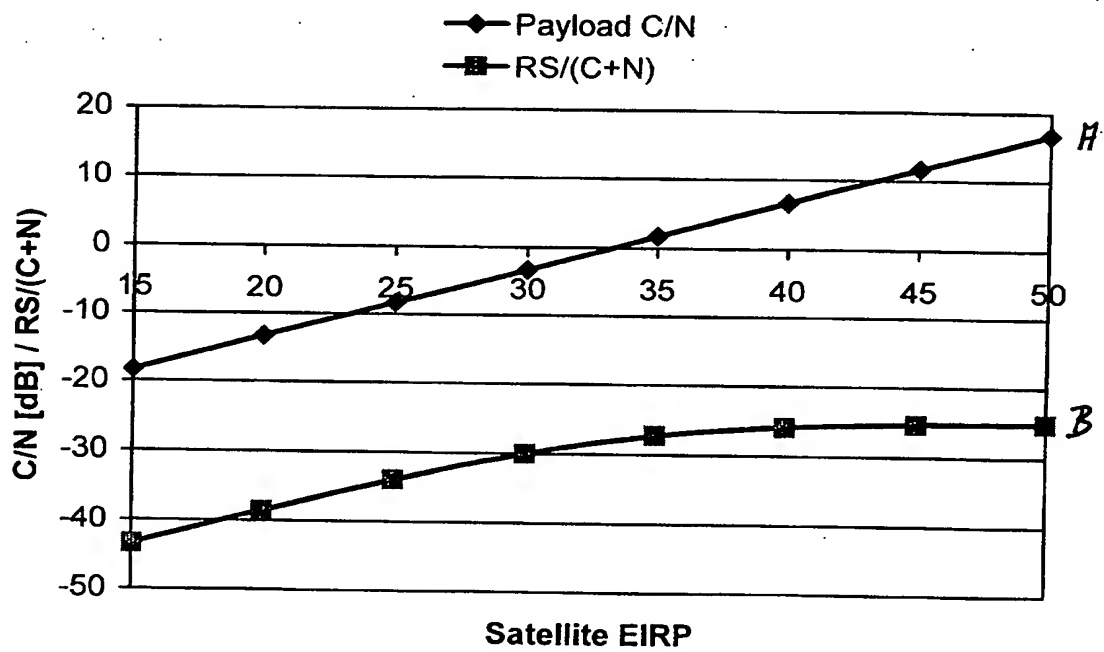


Fig. 12

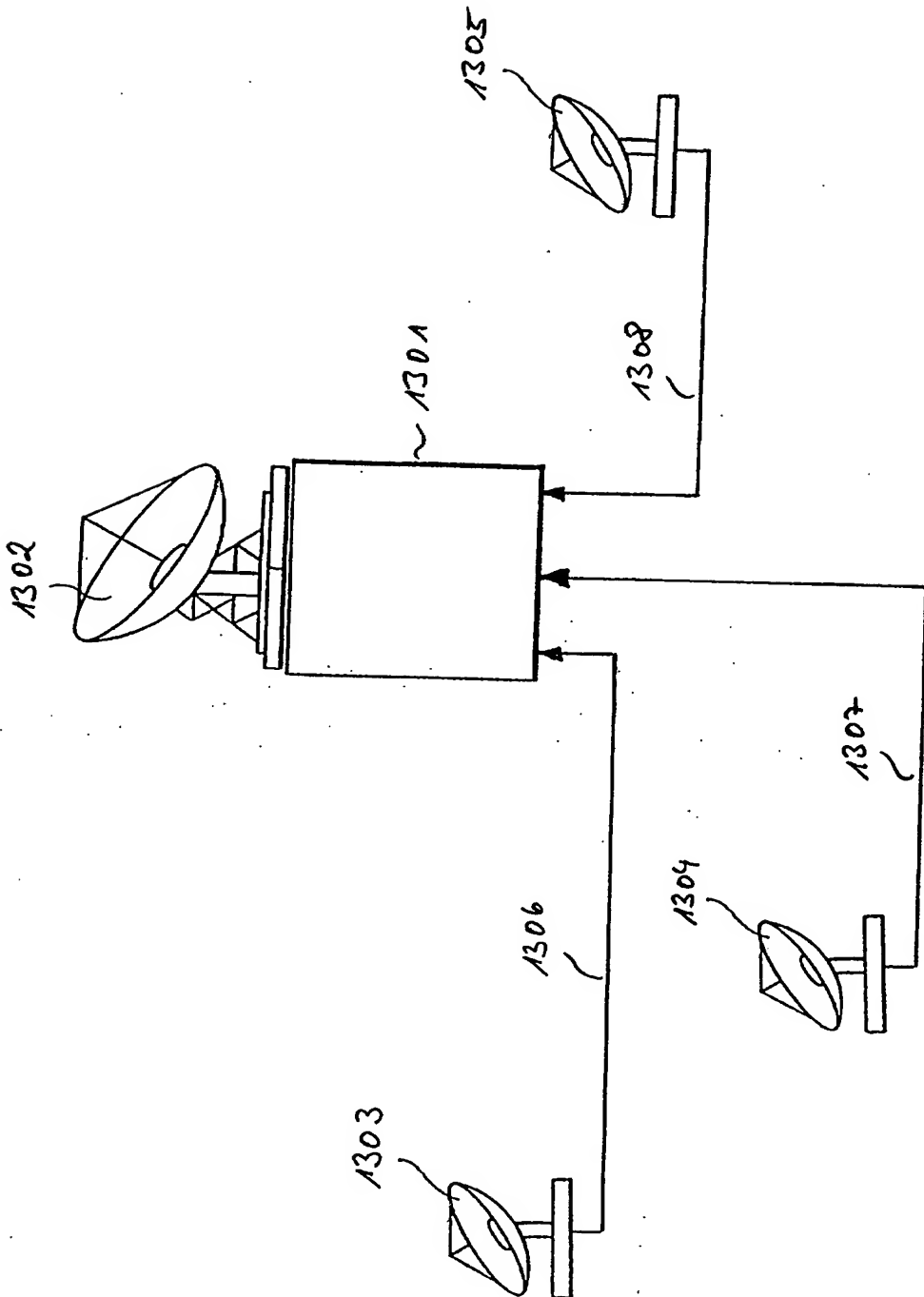


Fig. 13

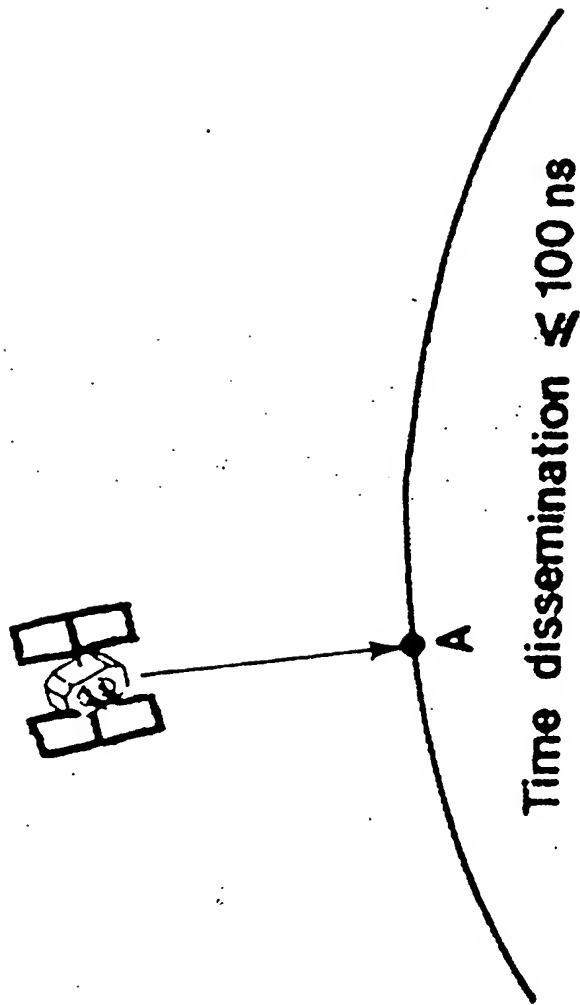


Fig. 14

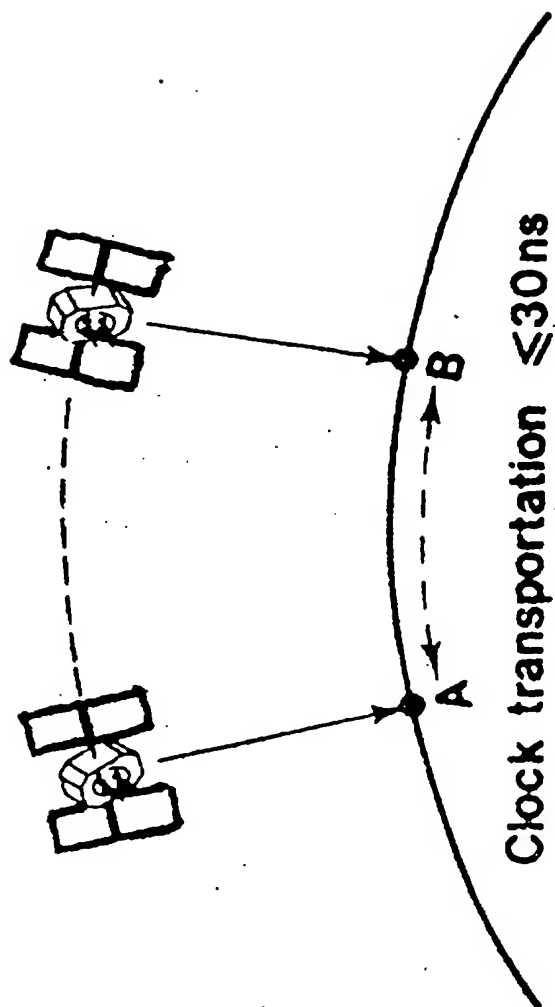


Fig. 15

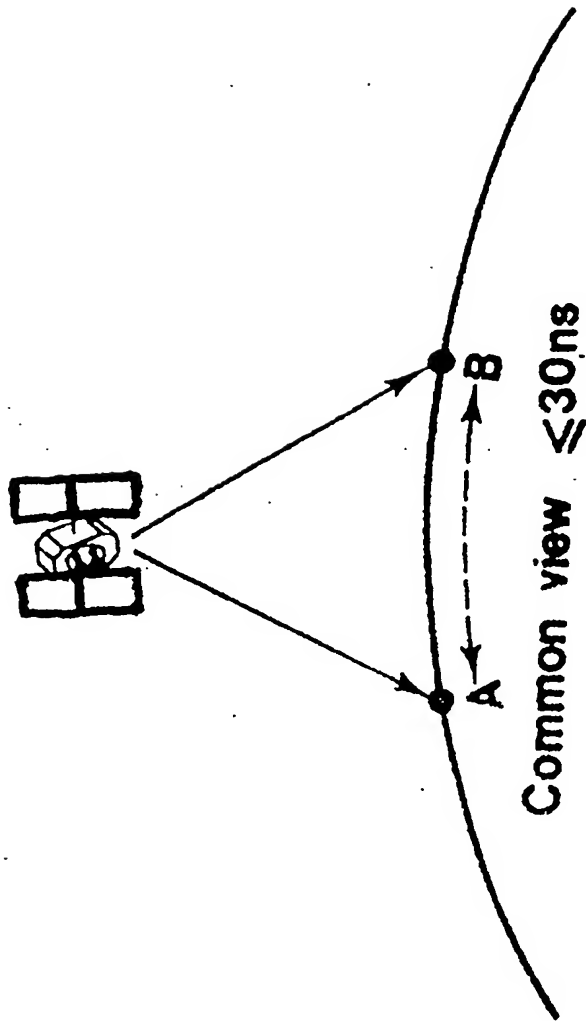


Fig. 16

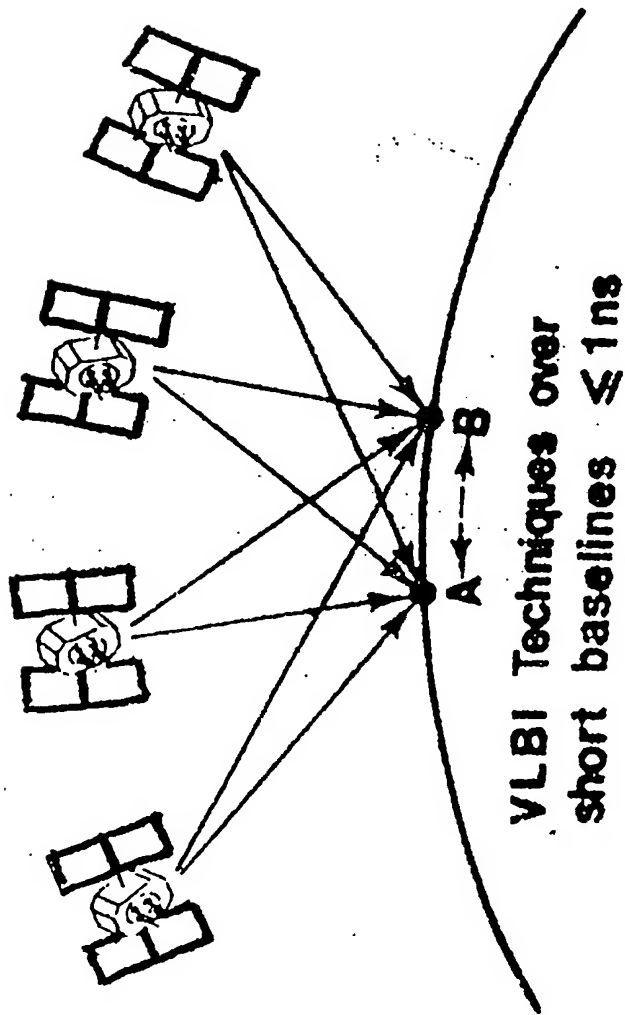


Fig. 17

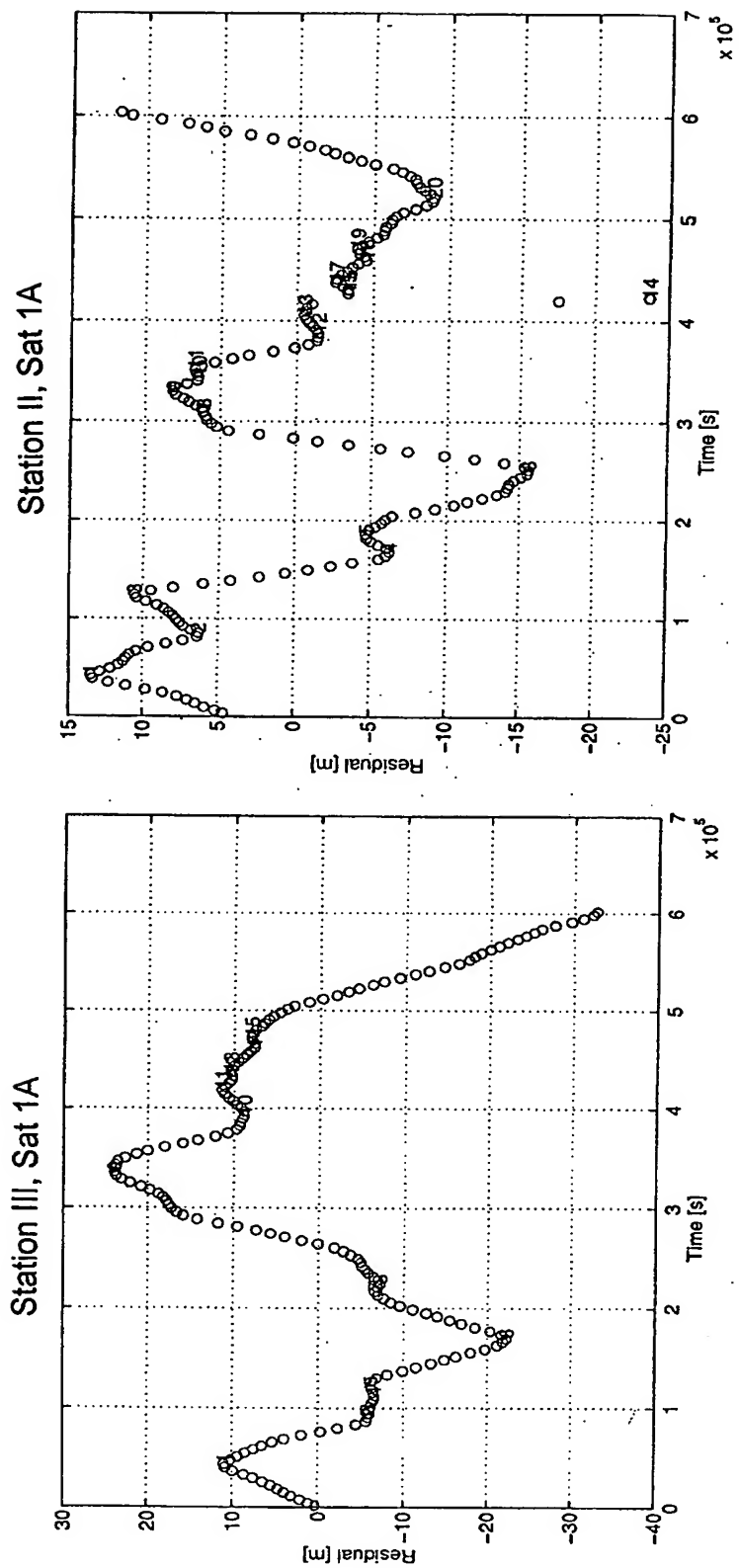


Fig. 18

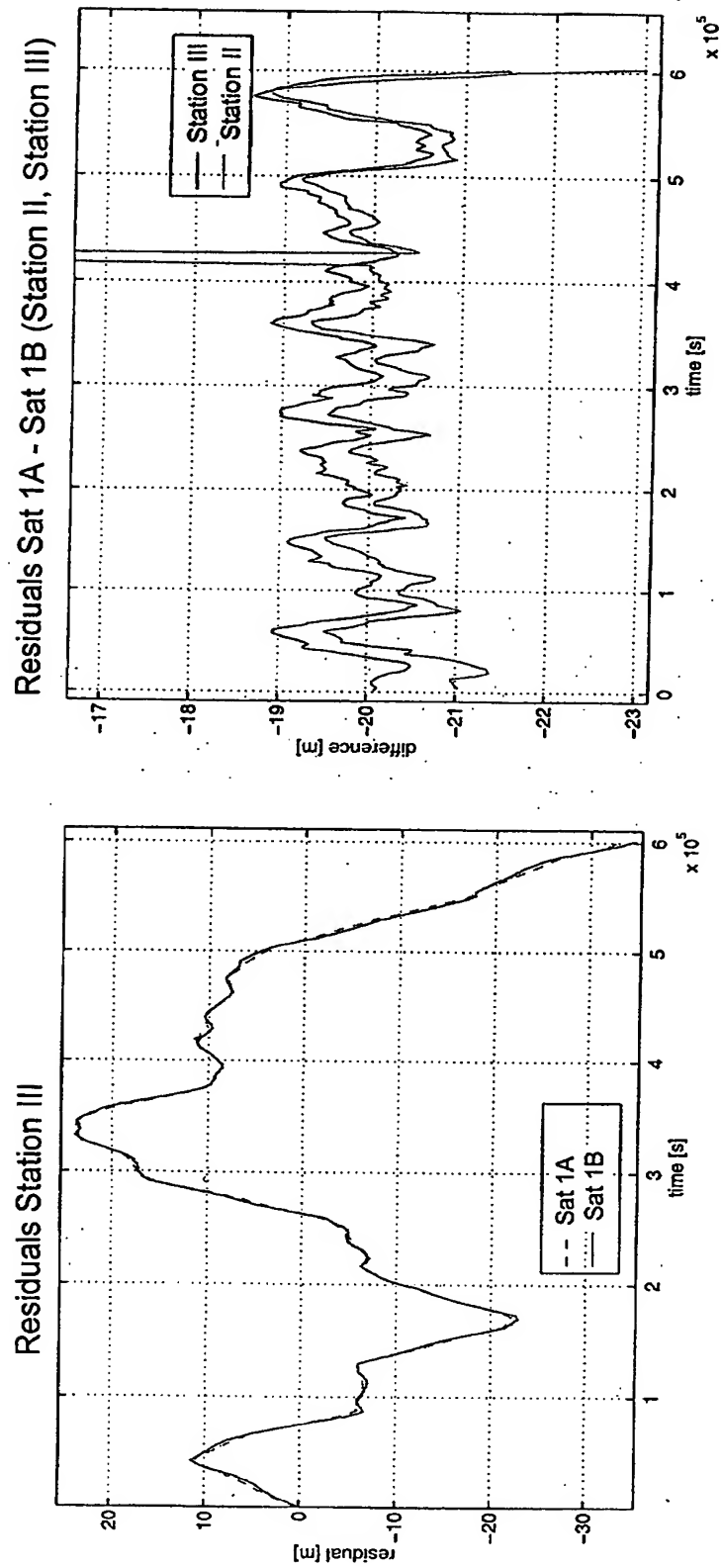


Fig. 19

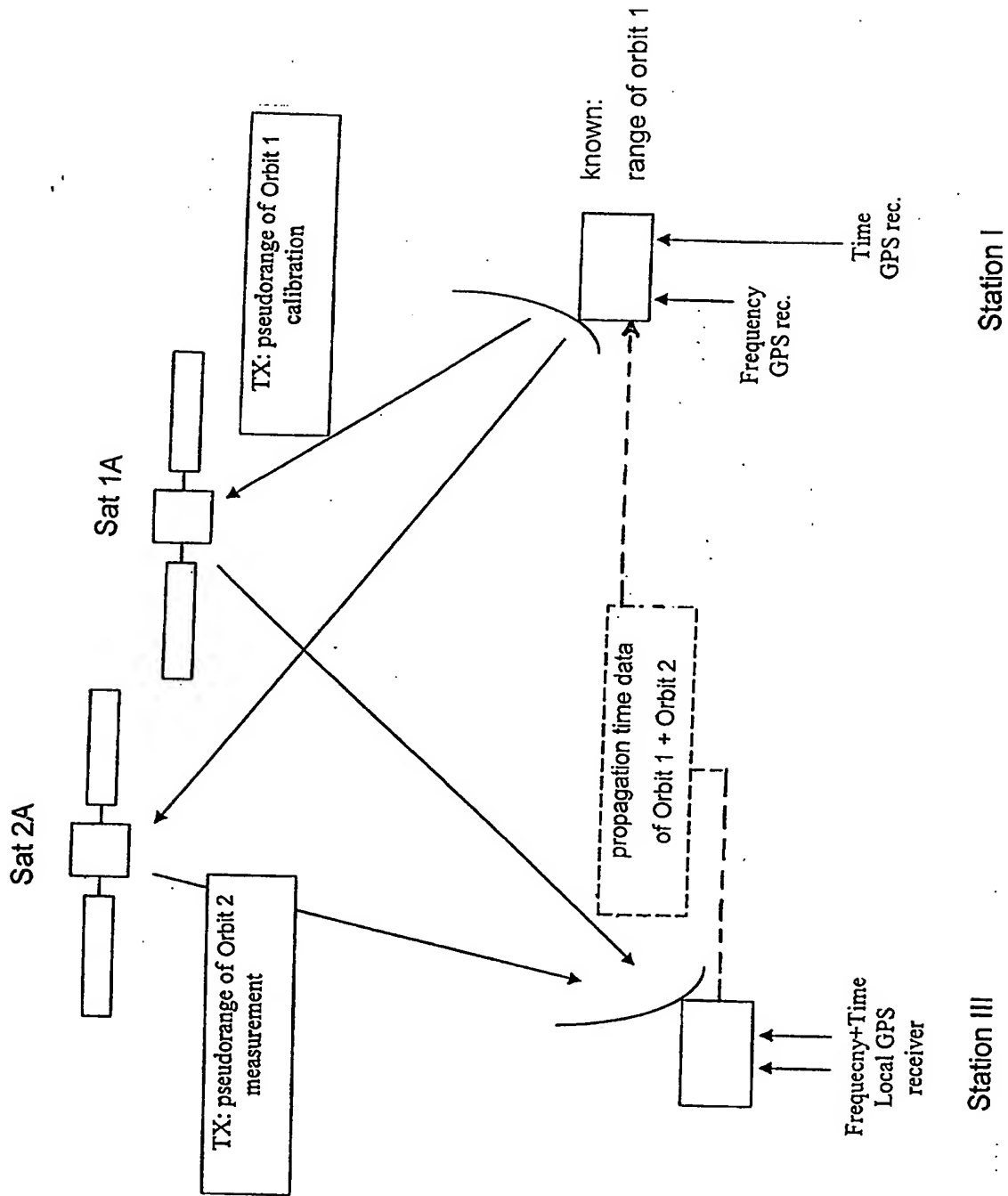


Fig. 20

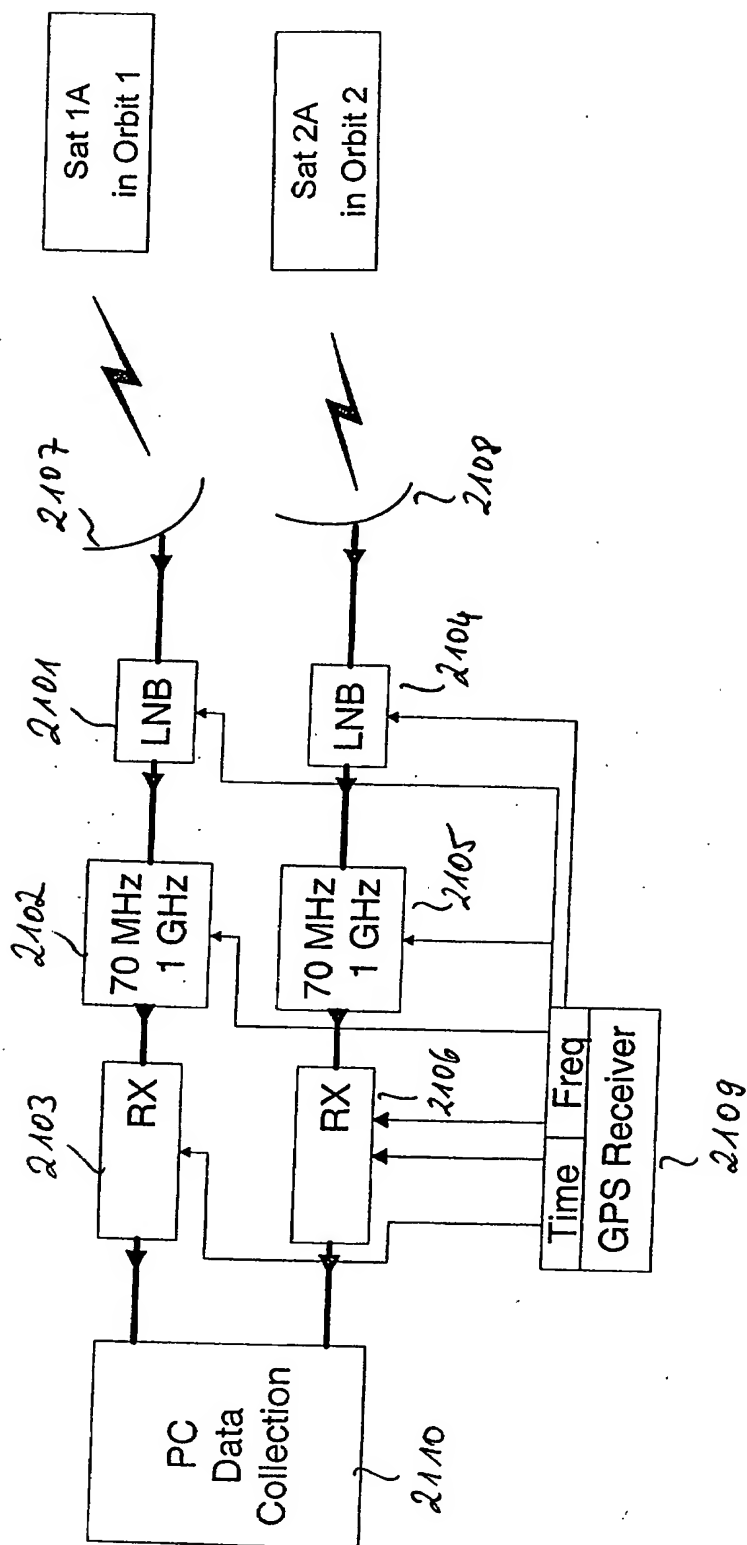


Fig. 21